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Direction Cosine Matrix

Highlights

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Discovering Thoughts, Inventing Future



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Implementation and Experimental Validation of a Real-Time Discontinuous PWM Technique for VSI Fed IM Drives

By P. Senthilkumar & N. P. Subramaniam

PRIST University, India

Abstract- Adjustable speed drive system requires variable voltage and frequency supply which is invariably obtained from a three-phase voltage source inverter (VSI). A number of Pulse Width Modulation (PWM) schemes are used to obtain variable voltage and frequency supply from an inverter. The most widely used PWM schemes for a three-phase VSI are carrier-based sinusoidal PWM and space vector PWM (SVPWM). This paper develops discontinuous PWM (DPWM) techniques for a voltage source inverter (VSI). Performance is evaluated in terms of total harmonic distortion (THD) in the output phase voltages. A significant reduction in switching losses is observed by adopting the proposed PWM schemes. The simulation and experimental results are provided to validate the concept.

Keywords: space vector, PWM, voltage source inverter, discontinuous PWM.

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Implementation and Experimental Validation of a Real-Time Discontinuous PWM Technique for VSI Fed IM Drives

P. Senthilkumar ^{α} & N. P. Subramaniam ^{σ}

Abstract- Adjustable speed drive system requires variable voltage and frequency supply which is invariably obtained from a three-phase voltage source inverter (VSI). A number of Pulse Width Modulation (PWM) schemes are used to obtain variable voltage and frequency supply from an inverter. The most widely used PWM schemes for a three-phase VSI are carrier-based sinusoidal PWM and space vector PWM (SVPWM). This paper develops discontinuous PWM (DPWM) techniques for a voltage source inverter (VSI). Performance is evaluated in terms of total harmonic distortion (THD) in the output phase voltages. A significant reduction in switching losses is observed by adopting the proposed PWM schemes. The simulation and experimental results are provided to validate the concept.

Keywords: space vector, PWM, voltage source inverter, discontinuous PWM.

I. INTRODUCTION

hree phase DC/AC Voltage Source Inverters (VSI's) are now used extensively in motor drives, active filters and unified power flow controllers in power systems and uninterrupted power supplies to generate controllable frequency and AC voltage magnitudes using various pulse-width modulation (PWM) strategies. Of the possible PWM methods, the carrier-based PWM is very popular due to its simplicity of implementation, defined harmonic waveform characteristics and low harmonic distortion. Two main implementation techniques exist for PWM: the direct digital method and the sine-triangle intersection scheme. In the traditional sine-triangle intersection PWM (SPWM) technique, three reference modulation signals are compared with a triangular carrier signal and the intersections define the switching instants of the controllable devices. PWM strategies are required for switching the devices in a VSI appropriately to generate variable voltage, variable frequency, 3-phase AC required for the variable speed induction motor drive. The performance of the drive strongly depends on the Pulse width Modulation (PWM) technique employed. At high power levels, the inverter distortion is guite high. Hence, PWM strategies for highpower drives must aim in reducing the inverter switching losses, harmonic distortion, subject to low switching frequencies of the inverter.

This zero sequence waveform is used to alter the duty cycle of the inverter switches. Adding the same zero sequence waveform to each of the three reference phase voltages does not change the inverter output lineline voltage per carrier cycle average value; however, if the waveform is properly selected, one can achieve any of the followings: switching losses can be drastically reduced, the waveform quality may be improved, the linear modulation range can be extended, and common mode voltage of motor drives can also be drastically diminished. These potentials have been explored leading to investigations into and determination of various zero sequence waveforms, resulting in a large number of published carrier-based PWM methods. When the augmenting zero sequence waveform is continuous, the continuous PWM (CPWM) scheme is produced; however when the zero sequence signal is discontinuous with a potential for the modulator to have phase segments clamped either to the positive or negative rails, the scheme is called the discontinuous PWM (DPWM).

The zero-voltage injection has been fully exploited but to different purposes: adding a zerovoltage sequence can serve not only to linearity extension, but also to reducing switching losses. In order to reduce switching losses the simplest method is not to switch: the idea was possible using zero-voltage sequences that saturate one of the modulation wave of the three phases.

There are about six variations reported in the literature under the name Discontinuous Pulse Width Modulation (DPWM) viz., DPWM0, DPMW1, DPWM2, DPWM3, DPWMMAX and DPWMMIN. The basic idea behind all these variations are employing discontinuous modulating signal instead of basic sinusoidal signal while carrier remains the triangular function. DPWM modulation techniques have the advantage of eliminating one switching transition in each half carrier interval, which allows the switching frequency to increase by a nominal value of 3/2 accordingly for the same inverter losses. This may improve the harmonic performance of the inverter by virtue of the reduced influence of higher frequency switching harmonics.

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The proposed carrier-based non-sinusoidal and discontinuous modulation schemes are experimentally implemented with an TMS320F2812 DSP Processor, intelligent power modules and used to modulate a three-phase inverter feeding a three-phase induction machine.

II. Principle of Dpwm and its Variants

a) 120° Discontinuous Modulation

In this type of modulation, each phase leg in turn is now continuously locked to the upper or lower DC rail (whichever is chosen) for one-third of the fundamental cycle (120°) .DPWMMAX and DPWMMIN strategies come under this category. In DPWMMAX, each phase around the peak is locked to the upper DC rail for one-third of the fundamental cycle (120°) as shown in Figure 1(a) and hence the name. Each phase in DPWMMIN, is locked to the lower DC rail for one-third of the fundamental cycle (120°) as shown in the Figure 1(b).Due to the asymmetry in line-to-line voltage, it has been expected that the harmonic performance of 120° discontinuous PWM will be suboptimal compared to continuous modulation strategies. However, it has been identified that to maintain the same effective phase lea switching frequency as for continuous modulation, the space vector calculation frequency (i.e. twice the carrier frequency) should be increased by approximately 50%, since each phase leg only switches during two-thirds of each fundamental cycle. It is trade-off between these two effects that offers potential advantages for 120° discontinuous PWM under some modulation conditions. A further limitation with this strategy is that one device of each phase leg is always turned off during its 120° unmodulated region, while the other device is always conducting. Hence conduction losses are not shared equally across the two devices in each phase leg.





However, this limitation can be avoided by clamping to alternate DC buses every half cycle (i.e. cycle between the DPWMIN and the DPWMMAX strategies).

b) 60° and 120° Discontinuous Modulation

Another improved discontinuous modulation strategy in which each phase leg is now unmodulated for only 60° at a time (alternately switched to the upper or to the lower DC rail).It has the particular benefit that

the line-to-line switched voltages are symmetrical. This type of discontinuous switching pattern is termed as DPWM1 and centers the non-switching periods for each phase leg symmetrically around the positive and negative peaks of its fundamental reference voltage. This is the best position for resistive load, since the peaks of the line currents follow the peaks of the fundamental voltages. Therefore each phase leg does not switch just when the current is at its maximum value, and this obviously minimizes switching losses. The modulating waveform for DPWM1 is as shown in the Figure 2(a).



Figure 2 : Modulating waveform for phase 'a' (a)DPWM1 (a) DPWM2

It is viable to place each 60° non-switching period anywhere within the 120° region where the appropriate phase leg reference voltage is the maximum/minimum of the three-phase set. For example, the $60^{\circ}\,\text{``clamp}$ to $\,+\,V_{dc}\,\text{``non}$ switching period for phase leg 'a' can be placed anywhere in the region $-\frac{\pi}{3} \leftrightarrow \frac{\pi}{3}$, since the 'a' phase reference voltage is more positive than the other two phase reference voltages during this period. Similarly, the "clamp to - V_{dc} " nonswitching period for phase leg 'a' can be placed anywhere in the region $\frac{2\pi}{3} \leftrightarrow \frac{4\pi}{3}$, since this is the region where the 'a' phase fundamental voltage is more negative than the other two-phase reference voltages. This freedom allows alternative 60° discontinuous PWM strategies to be considered which minimize the switching losses for loads which are not unity power factor.

For a lagging power factor (pf) load, it is clearly preferable to retard the non-switching period by up to a maximum of 30° (pf of 0.866 lag), depending on the load current power factor. This is the discontinuous modulation strategy DPWM2 and its modulating waveform is as shown in figure 2(ii).For a leading power factor load, the non switching period can be advanced by up to 30° (pf of 0.866 lead).This is discontinuous modulation strategy DPWM0 and its modulating waveform is shown in Figure 3(a). It should be noted however, that incorporating this advance/retard of the

non switching period reintroduces asymmetry into the line-to-line voltage as shown in the figure 3(b).

The final variation is to clamp the phase legs to the opposite DC rails in each 60° segment as shown in the figure. This strategy is referred as 30° discontinuous modulation-DPWM3, since it requires the clamping phase leg to change every 30° as shown in figure 3(ii).





In DPWM, the modulating signal is discontinuous in nature comprised of different functions for each section of time reference. There are six functions which are substituted in various sections of time reference. The differences among the DPWM schemes are just interchange of these sections. Table 1, 2, 3, 4, 5 and 6 details the expressions for the modulation waves of various DPWM variants for one complete cycle.

In all the cases, it has been understood that the phase leg reference segments are in fact sections of the required line-to-line voltage, referenced to the phase leg which is clamped to either the upper or the lower DC rail. The variation of M does not have any effect in the saturation region of the modulating wave.

Alpha (degree)	Phase A	Phase B	Phase C
0-30°	+1	$+1-\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	$+1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$
30°-90°	$-1-\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	$-1+\sqrt{3}*M*\sin(\theta)$	-1
90°-150°	$+1+\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	+1	$+1-\sqrt{3}*M*\sin(\theta)$
150º-210º	-1	$-1-\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	$-1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$
210º-270º	$+1-\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	$+1+\sqrt{3}*M*\sin(\theta)$	+1
270°-330°	$-1+\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	-1	$-1-\sqrt{3}*M*\sin(\theta)$
330°-360°	+1	$+1-\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	$+1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$

Table 1	Composite	modulating	function-	DPWM1
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	inposite m	oquiating it		

Alpha (degree)	Phase A	Phase B	Phase C
0-60°	+1	$+1-\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	$+1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$
60°-180°	$+1+\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	+1	$+1-\sqrt{3}*M*\sin(\theta)$
180º-300º	$+1-\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	$+1+\sqrt{3}*M*sin(\theta)$	+1
300°-360°	+1	$+1-\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	$+1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$

Alpha (degree)	a Phase A Phase B		Phase C	
0-60 ⁰	+1	$+1-\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	$+1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	
60°_120°	$-1-\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	$-1+\sqrt{3}*M*\sin(\theta)$	-1	
120º-180º	$+1+\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	+1	$+1-\sqrt{3}*M*\sin(\theta)$	
180°-240°	-1	$-1-\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	$-1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	
240°-300°	$+1-\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	$+1+\sqrt{3}*M*\sin(\theta)$	+1	
300°-360°	$-1+\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	-1	$-1-\sqrt{3}*M*\sin(\theta)$	

Table 3 : Composite modulating function- DPWM2

Table 4 : Composite modulating function- DPWM0

Alpha (degree)	Phase A	Phase B	Phase C
0-60 ⁰	$-1-\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	$-1+\sqrt{3}*M*\sin(\theta)$	-1
60 ⁰ -120 ⁰	$+1+\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	+1	$+1-\sqrt{3}*M*\sin(\theta)$
120º-180º	-1	$-1-\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	$-1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$
180º-240º	$+1-\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	$+1+\sqrt{3}*M*\sin(\theta)$	+1
240º-300º	$-1+\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	-1	$-1-\sqrt{3}*M*\sin(\theta)$
300°-360°	+1	$+1-\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	$+1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$

Table 5 : Composite modulating function- DPWMMIN

Alpha (degree)	Phase A	Phase B	Phase C
0-120°	$-1-\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	$-1+\sqrt{3}*M*sin(\theta)$	-1
120º-240º	-1	$-1-\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	$-1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$
240º-360º	$-1+\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	-1	$-1-\sqrt{3}*M*\sin(\theta)$

Alpha (degree)	Phase A	Phase B	Phase C
0-30°	$-1-\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	$-1+\sqrt{3}*M*\sin(\theta)$	-1
30°_60°	+1	$+1-\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	$+1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$
60°-90°	$-1+\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	+1	$+1-\sqrt{3}*M*\sin(\theta)$
90°-120°	$-1-\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	$-1+\sqrt{3}*M*\sin(\theta)$	-1
120º-150º	-1	$-1-\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	$-1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$
150º-180º	$+1+\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	+1	$+1-\sqrt{3}*M*\sin(\theta)$
180º-210º	$+1-\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	$+1+\sqrt{3}*M*\sin(\theta)$	+1
210º-240º	-1	$-1-\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	$-1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$
240º-270º	$-1+\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	-1	$-1-\sqrt{3}*M*\sin(\theta)$
270º-300º	$+1-\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$	$+1+\sqrt{3}*M*\sin(\theta)$	+1
300°-330°	+1	+1- $\sqrt{3}$ *M*cos(θ + $\frac{\pi}{6}$)	$+1+\sqrt{3}*M*\cos(\theta+\frac{5\pi}{6})$
330°-360°	$-1+\sqrt{3}*M*\cos(\theta+\frac{\pi}{6})$	-1	$-1-\sqrt{3}*M*\sin(\theta)$

Table 6 : Composite	modulating	function-	DPWM3
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III. Performance Evaluation

Parameters that is taken in consideration for performance evaluation of the proposed DPWM methods namely Total Harmonic Distortion (THD) is defined in by equations (1),

THD =
$$\sqrt{\sum_{n=3,5,7..}^{\infty} \left(\frac{V_n}{V_1}\right)^2}$$
 (1)

Where V_n represents n^{th} order harmonic component and V_1 represent fundamental output phase voltages. The lowest order harmonic contents (upto 25^{th} order) are considered for calculation of THD. The resulting THD are shown in Figure 4. Their measured %THDs is compared (Table-7).

Implementation and Experimental Validation of a Real-Time Discontinuous PWM Technique for VSI Fed IM Drives



Figure 4 : Harmonic spectra (% of fundamental) , (a) for DPWMMAX ; (b) for DPMIN ; (c) for DPWM 1; (d) for DPWM 2 ; (e) for DPWM 3 and (f) for DPWM 0

	M=0.8		M=0.6		M=0.3		
	at 30Hz	at 50Hz	at 30Hz	at 50Hz	at 30Hz	at 50Hz	
DPWMMAX	4.21	5.21	6.70	8.10	20.01	21.46	
DPWMMIN	4.21	5.20	6.71	8.10	20.10	22.60	
DPWM1	3.65	5.41	3.99	11.20	11.07	21.93	
DPWM2	3.61	5.40	4.20	9.52	11.31	19.62	
DPWM3	3.70	5.02	4.48	9.40	10.42	19.70	
DPWM0	3.71	5.01	4.54	9.42	10.46	19.68	

Table 7 : Comparison of % THDs of various PWM methods at different modulation indices (M)

It is seen from that at low modulation index the THD of DPWMMAX is lowest compared to all other schemes and DPWM1 has highest THD. Since the DPWMMAX may not be employed for implementation purposes, thus the next best scheme is DPWM2. At high modulation index the value of THD is smaller compared to the value at lower modulation index, which implies that the output voltage is very near to the sinusoidal. DPWM1 has the lowest THD and DPWM0 and DPWM3 has highest THD at high modulation index.

IV. EXPERIMENTAL RESULTS

The proposed DPWM techniques has been implemented in real time using Texas Instrument

TMS320F2812 DSP Processor, intelligent power modules and was programmed through the software Vissim. The DSP board is integrated interfacing communication board. The PC is connected to the DSP board through printer parallel port.

The DSP board is connected through cable to the Inverter intelligent power modules. Current sensors are used for feedback purposes. The code is run using Code Composer studio CCS 3.2V, and the .out file thus created is then converted to the ASCII file which is loaded to the DSP for further processing. The complete experimental set up is illustrated in Figure 5 (a) and (b).



Figure 5 : Experimental set-up (a) TMS320F2812 DSP board and Intelligent Power Module (b) Induction Motor with Load setup

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Implementation and Experimental Validation of a Real-Time Discontinuous PWM Technique for VSI Fed IM Drives



Figure 6 : Line voltage and Line Current of, (a) for DPWMMAX; (b) for DPMIN; (c) for DPWM 1; (d) for DPWM 2; (e) for DPWM 3 and (f) for DPWM 0

Illustrative experimental results are given in Figures 6 (a), (b), (c), (d), (e), (f) and (g) showing voltage and current waveforms when a loaded induction machine is fed with a voltage source inverter under various types of generalized discontinuous modulation.

v. Conclusions

The paper present PWM technique termed as Discontinuous PWM for voltage source inverter. This

proposed method of PWM offers a reduction in overall number of switching and consequently switching losses.. The analysis is done on the basis of THD in output phase voltages. It can be concluded that the DPWMMAX provide lowest THD for low modulation index. However, this method is not recommended for practical implementation as this may shorten the life of inverter. DPWM2 offers the next best result and thus it may be used for implementation. At high modulation index DPWM1 is recommended for use. The viability of the proposed schemes is validated using simulation and experimental result

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Transmission Line Fault Detection Using Android Application Via Bluetooth

By MD Asaduzzaman Nur, Jahidul Islam, Md. Golam Mostofa & Moshiul Alam Chowdhury

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Abstract- Technological advancement and its incorporation is playing significant in human life. In present days, the demand on the electric power for the household, commercial and industrial loads is increasing. Also, the management of electric power distribution system is becoming more complex. Bluetooth based fault detection is a newly developing concept in the power system fault detection. This is a part of smart grid. The system is designed to detect the transmission line fault for the user to easily recognize the current condition of the distribution line. The ultimate objective is to monitor the distribution line status continuously and hence to guard the fault of distribution line due to the constraints such as overvoltage, under voltage, SLG, DLG faults. If any of these does occurs then a user can easily detect the fault.

Keywords: microcontroller, transformer, ICE, bluetooth module, bridge rectifier, android application.

GJRE-F Classification : FOR Code: 090609



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Transmission Line Fault Detection Using Android Application Via Bluetooth

MD Asaduzzaman Nur $^{\alpha}$, Jahidul Islam $^{\sigma}$, Md.Golam Mostofa $^{\rho}$ & Moshiul Alam Chowdhury $^{\omega}$

Abstract- Technological advancement and its incorporation is playing significant in human life. In present days, the demand on the electric power for the household, commercial and industrial loads is increasing. Also, the management of electric power distribution system is becoming more complex. Bluetooth based fault detection is a newly developing concept in the power system fault detection. This is a part of smart grid. The system is designed to detect the transmission line fault for the user to easily recognize the current condition of the distribution line. The ultimate objective is to monitor the distribution line status continuously and hence to guard the fault of distribution line due to the constraints such as overvoltage, under voltage, SLG, DLG faults. If any of these does occurs then a user can easily detect the fault.

Keywords: microcontroller, transformer, ICE, bluetooth module, bridge rectifier, android application.

I. INTRODUCTION

Protection system's main function is to clear faults from the power system at high speed to ensure safety, minimize equipment damage and maintain power system stability[2]. Protection of power systems requires an understanding of system faults, their detection, and safe isolation of the faulted device. By taking an inventory of all the essential electrical loads and doing a basic electrical load evaluation [2], an idea regarding how much power our system needs to produce has been obtained. We are also aware about the power fluctuation situations also that means what voltage minimum / maximum we are getting from the A.C supply mains. In doing this project we would be using concepts of microcontrollers, Bluetooth Module and Android Application.

II. FAULT TYPES AND PROTECTION

a) Single-Line-to-Ground Fault

A short circuit between one line and ground, very often caused by physical contact, for example due to lightning or other commomeans. The single line to ground fault can occur in any of the three phases[1]. However, it is sufficient to analyze only one of the cases.

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Figure 1 : Single Line to Ground fault

b) Line-to-Line Fault

A short circuit between lines, caused by ionization of air, or when lines come into physical contact[1], for example due to a broken insulator. For a Line-to-line fault, the currents will be high, because the fault current is only limited by the inherent (natural) series impedance of the power system up to the point of faulty (refer Ohms law).





Double- Line -to Ground Fault

Two lines come into contact with the ground (and each other), also commonly due to stormy weather or some other means [1].



Figure 3 : Double line to Ground Fault

d) Lower Voltage

Low voltage is a relative term, the definition varying by context. Different definitions are used in transmission and distribution line, and in the electronics industry. Electrical safety codes define "low voltage" circuits that are exempt from the protection required at higher voltages. These definitions vary by country and specific code. Lower voltage is defined as incoming line voltage at the point of use which is smaller than the Public Service Commission's mandated legal limits; and/or smaller than the voltage ratings of the connected equipment. Lower voltage is considered a safety hazard by all industry standards and can cause premature failure of connected equipment. Devices could be damaged by lower line voltage.

e) Over Voltage

Overvoltage is defined as incoming line voltage at the point of use which is greater than the Public Service Commission's mandated legal limits; and/or greater than the voltage ratings of the connected equipment. Overvoltage is considered a safety hazard by all industry standards, and can cause premature failure of connected equipment. Overvoltage has been a widely known industry problem for many years, but it is not generally understood by many who have to deal with it. Power companies have been unable to control it adequately. Overvoltage occurs most often during severe cold winter weather for the following reasons: (1) Inadequate size of power distribution systems: (2) slow reaction time for power company's distribution systems to regulate voltage during extreme load variations; and (3) abrupt reductions of loads.

III. BLOCK DIAGRAM ARRANGEMENT

In figure 4, the basic arrangement of the implemented project can be found.



Figure 4 : Block diagram arrangement of the project

Among the major components required to establish the project, few of them are the power transformers (step down), microcontroller ATMEGA 16 and Bluetooth module ,Bridge rectifier.

a) Transformer

Transformer is an electrical device used to step up and down the AC voltages. There are two types

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of transformers: Step up and step down transformer. Step up transformer increases the magnitude of voltage while step down transformer decrease the magnitude of voltage. Depending on the ratio of the number of turns in the primary & secondary winding a transformer is characterized as step up or step down. For this project purpose, considering 1 Φ voltage to be around 220 V, 3 step down power transformer of rating 220/12 has been used to represent a realistic representation of the 3 Φ system.





b) Microcontroller ATMEGA 16

Brain of this project is Atmega16 microcontroller. It is a 8 bit Micro controller with RISC architecture. Its speed is up to 16MIPS throughput at 16MHz. It has 16K bytes of flash and 512bytes EEPROM. Operating voltage 2.7v -5.5v, in active mode it consumes only 1.1mA & in sleep mode it consumes less than 1uA current which made it a perfect choice for his project [3].



Figure 6 : Atmega16 microcontroller

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c) Bluetooth module

HC-06 has been used as Bluetooth module. The Baud rate is 9600. Master and slave mode can't be switched in this Module. HC-06 module have paired memory to remember last slave device. The working voltage is 3.3V, but it can work at 3.00-4.2v.The Current pairing 20~30mA, connected 8mA[5].



Figure 7: Bluetooth module HC-06

d) Bridge rectifier

An AC power source is required for powering major appliances but almost all electronic circuits require a steady DC supply. A simple rectifier circuit described in this project converts the input from AC source to DC voltage. Firstly, the AC input from mains is stepped down to a lower value of voltage. This AC supply is then passed through a rectifier circuit to remove the negative cycle of AC waveform. The resulting signal is then filtered to get the DC output. The major part of the circuit is connected to the secondary coil of the transformer which is comprised by diodes and capacitor. While the diodes act as a rectifiers, capacitor filters out the DC component from the circuit.



Figure 8 : Bridge rectifier

IV. SIMULATION

The initial stage, the circuits have been designed and simulated in PROTEUS. The circuit have been utilized to analyze the line to line, single line to ground, double line to ground, open conductor, over voltage and under voltage in the sides of the transformer. The circuit diagram can be found in figure 9.



Figure 9 : Circuit Simulation

V. HARDWARE IMPLEMENTATION

In reference to figure 4, the transmitting and receiving side can be described as follows:

a) Transmitting side

Heart of the project is the microcontroller ATMEGA 16. In general the normal distribution phase voltage is 220 V, in this project we used a step down transformer 220/12 V for converting the phase voltage from 220 V to 12 V. Then, a bridge rectifier has been used for converting the 12 V ac to 12 V dc; after that, applied voltage divider converts the 12 V to 5 V because the microcontroller works at maximum 5 V. By this process the three distribution phase is connected into three microcontroller pins. At cases, when the distribution side is in load shedding protection of transformer must be ensured, which is why the microcontroller power is given from an external power source (5 V battery) backup and also the Bluetooth module power is given from external power source(4V battery). Bluetooth module communicates with atmeg16 through UART. RXD of Bluetooth module is connected with TXD of atmega16 and TXD of Bluetooth module is connected to RXD of atmega16.

b) Receiving side

In receiver Side an Android Phone is connected with Transmitting side via Bluetooth Module CI Android Apps[6]. At first we connect the android phone with HC- 06 and the password of the HC-06 is 1234 then android apps shows the data which is send by transmitting side. The communication protocol is UART and baud rate is 9600[4].

In view of the descriptions above, the implemented hardware can be found in figure 10.



Figure 10 : Implemented Hardware model



Figure 11 : LCD status indicator

As viewed from figure 10, the system was found to be balanced three phase system. In figure 11, the corresponding representation appears in the LCD display with the phase voltages in all the phases to be around 219 V. The view from Android Phone is also shown in figure 12.



Figure 12 : Android Phone output

VI. FUTURE PROSPECTS

In view of a wide and Short range of possibilities on the basis of Bluetooth based fault Detection system, a few has been depicted below:

- 1. Fault detection of a Generator and motor.
- 2. Improvements to human-machine interface.
- 3. Improvements in computer-based protection of Industry automation.

VII. CONCLUSION

Microcontroller and Bluetooth based fault detection system is a reliable technique for monitoring and controlling the electric distribution system, the microcontroller works up to 100 °C temperature. For Short distance data transmission Bluetooth technology is a reliable and robust one. Any kind of fault occurring in the distribution system results the Bluetooth modules to send instant messages automatically to the nearest user. Bluetooth based microcontroller Fault detection system will serve as a reliable, easy and cost effective solution for monitoring and controlling the electric distribution system.

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Optimal Power Flow in the Presence of Practical Constraints and with TCSC using Imperialistic Competitive Algorithm

By G. Nageswara Reddy, S. S. Dash, S. Sivangaraju & Ch. V. Suresh

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Abstract- At present, power system operation, control, Management becomes very complex due to continuously increasing demand. Flexible AC Transmission System (FACTS) controllers are used to increase power transfer capability of the transmission lines closer to their limits. This paper proposed a methodology to solve Optimal Power Flow (OPF) problem in the presence of Thyristor Controlled Series Capacitor (TCSC) while satisfying system operating and practical constraints. A novel cost objective function is formulated by combining investment cost of the TCSC with the conventional fuel cost function. The proposed methodology is tested on standard IEEE-14 bus test system with supporting results.

Keywords: economic dispatch, ramp-rate limit, prohibited operating zones, TCSC, investment cost.

GJRE-F Classification : FOR Code: 090607



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Optimal Power Flow in the Presence of Practical Constraints and with TCSC using Imperialistic Competitive Algorithm

G. Nageswara Reddy ^a, S. S. Dash ^a, S. Sivangaraju ^a & Ch. V. Suresh ^a

Abstract- At present, power system operation, control, Management becomes very complex due to continuously increasing demand. Flexible AC Transmission System (FACTS) controllers are used to increase power transfer capability of the transmission lines closer to their limits. This paper proposed a methodology to solve Optimal Power Flow (OPF) problem in the presence of Thyristor Controlled Series Capacitor (TCSC) while satisfying system operating and practical constraints. A novel cost objective function is formulated by combining investment cost of the TCSC with the conventional fuel cost function. The proposed methodology is tested on standard IEEE-14 bus test system with supporting results.

Keywords: economic dispatch, ramp-rate limit, prohibited operating zones, TCSC, investment cost.

I. INTRODUCTION

important problem ost in power systemoperation and control is Economic Load(ED). This problem should be optimized tomaximize the lowest cost of fuel to meet the load. The main objective of OPF problem is to optimize the chosen objectives such as classical quadratic fuel cost function and non-convex fuel cost functions in the presence of operating as well as practical constraints. In general, OPF problem is described as a highly constrained, large-scale, non-linear, non-convex optimization problem.

In the last three decades eminent population based optimization techniques have been developed. They are: Grenade Explosion Method (GEM) [1], Ant Colony Optimization (ACO) [2], Shuffled Frog Leaping (SFL) [3], Artificial Immune Algorithms (AIA) [4], Harmony Search (HS) [5], Genetic Algorithm (GA) [6], Artificial Bee Colony (ABC) [7], Particle Swarm Optimization (PSO) [8], Bacteria Foraging Optimization (BFO) [9], Gravitational Search Algorithm

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(GSA) [10], Biogeography-Based Optimization (BBO) [11] and Differential Evolution (DE) [12] these algorithms have been applied to solve OPF problems effectively for specific type of problems.

All of these algorithms are population based optimization methods, but they have some limitations in some aspects. To overcome this limitations research is continued, to improve the performance of the above methods. Improvement of the existing methods is reported in GA [13], PSO [14, 15], ACO [16, 17], ABC [18, 19]. Combining merits in the different optimization techniques known as hybridization of the optimization techniques. Some hybrid algorithms can found in [20, 21], etc.

The performance comparison of optimization techniques such as DE, PSO and GA for power loss minimization using FACTS devices is presented [22]. Optimal location and best setting of TCSC is proposed. Best location is performed using sensitivity analysis [23]. In [24], the improvement of the power system security against contingencies with TCSC approach is proposed. In this GA and PSO algorithms are used to find the optimal location and parameter settings of TCSC under contingency. Mathematical modeling of varies series FACTS devices are presented in economic dispatch problems [25]. Here selection of optimal location of FACTS devices is based on the improvement of the economic dispatch.

In [26], OPF problem formulation has been presented along with considering effects of FACTS devices and power flow constraints. To solve OPF problem a two stage problem formulation has been proposed. This reference [27], Presents the performance comparison of PSO and GA optimization techniques for FACTS based controller design. In [28], the main aim is the optimal allocation of TCSC device in the power system so the power loss become minimized and also increase power transfer capacity of the transmission line that ultimately yields minimum operating cost with different load conditions. In this, first the locations of the TCSC device is identified by calculating line flows. TCSC is placed in line where reactive power flows are very high. To solve OPF problem Hybrid GA incorporating TCSC is proposed. To select best control variables, to minimize the generation cost and maintain the power flow within the control

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range, hybrid GA is integrated with conventional OPF [29].

The generation cost is minimized subjected to equality and inequality constrains. The optimal active power flow control strategy with TCSC is proposed. In this method problem is solved in two steps. First step is power flow control and second step is normal OPF problem [30]. In [31], presents solution of OPF with different objective functions such as generation cost and total power loss minimization using GA. In base case OPF solution is obtained with generation cost minimization and optimal setting of the control parameters are determined. In this OPF with TCSC is carried out with generation cost and total power loss as objective functions.

This paper mainly concentrates on certain power system issues such as quadratic fuel cost function and non-convex fuel cost functions are optimized while satisfying operating and practical constraints is analyzed in the presence of TCSC. A methodology to install TCSC in an optimal location is presented. The proposed OPF problem is solved using Imperialistic Competitive Algorithm (ICA) while satisfying equality, in-equality, ramp-rate constraints and prohibited operating zones in the presence of TCSC.

II. MODELING OF TCSC

TCSC is a variable impedance type series compensator which is connected in series with the transmission line to increase the power transfer capability, improve transient stability, reduces transmission losses. The basic configuration of TCSC and its equivalent pi-representation for transmission line modeling is shown in Fig.1.

a) Basic model of TCSC



b) Equivalent pi-representation of TCSC



Fig 1 : TCSC model

This device is incorporated in a line between buses, 's' and 'r' then, the new line reactance can be expressed as

X sr new = X sr - XC (1)

Therefore new line admittance between buses's' and 'r' can be derived as follows

$$Y'_{sr} = \frac{1}{Z'_{sr}} = \frac{1}{R_{sr} + j(X_{sr} - X_C)}$$
 (2)

$$Y_{sr}^{'} = G_{sr}^{'} + jB_{sr}^{'} = \frac{R_{sr} - j(X_{sr} - X_{C})}{R_{sr}^{2} + (X_{sr} - X_{C})^{2}}$$
(3)

$$G'_{sr} = \frac{R_{sr}}{R_{sr}^2 + (X_{sr} - X_C)^2}$$
; $B'_{sr} = -\frac{(X_{sr} - X_C)}{R_{sr}^2 + (X_{sr} - X_C)^2}$

The modified active and reactive power flows from bus-s to bus-r, and from bus-r to bus-s of a line having series impedance and a series reactance are

$$P_{sr}^{TCSC} = V_s^2 G'_{sr} - V_s V_r \left(G'_{sr} \cos(\delta_{sr}) + B'_{sr} \sin(\delta_{sr}) \right)$$

$$Q_{sr}^{TCSC} = -V_s^2 \left(B'_{sr} + B_{sh} \right) - V_s V_r \left(G'_{sr} \sin(\delta_{sr}) - B'_{sr} \cos(\delta_{sr}) \right)$$

$$P_{rs}^{TCSC} = V_r^2 G'_{sr} - V_s V_r \left(G'_{sr} \cos(\delta_{sr}) - B'_{sr} \sin(\delta_{sr}) \right)$$

$$Q_{sr}^{TCSC} = -V_r^2 \left(B'_{sr} + B_{sh} \right) + V_s V_r \left(G'_{sr} \sin(\delta_{sr}) + B'_{sr} \cos(\delta_{sr}) \right)$$

The power loss in the line with TCSC can be written as

$$P_{Loss} = P_{sr}^{TCSC} + P_{rs}^{TCSC} = G'_{sr} \left(V_s^2 + V_r^2 \right) - 2V_s V_r G'_{sr} Cos \left(\delta_{sr} \right)$$

$$Q_{Loss} = Q_{sr}^{TCSC} + Q_{rs}^{TCSC} = -\left(V_s^2 + V_r^2 \right) \left(B'_{sr} + B_{sh} \right) + 2V_s V_r B'_{sr} Cos \left(\delta_{sr} \right)$$

$$V_{s} \xrightarrow{P_{s}^{TCSC}} Z_{sr} = r_{sr} + jx_{sr} \xrightarrow{P_{r}^{TCSC}} V_{r}$$



Due to TCSC, the change in line flow can be represented as a line without TCSC plus with power injected at the sending and receiving ends of the line with device as shown in Fig.2. The active and reactive power injections at bus-s and bus-r can be written as

$$P_s^{TCSC} = V_s^2 \Delta G_{sr} - V_s V_r \left[\Delta G_{sr} \cos(\delta_{sr}) + \Delta B_{sr} \sin(\delta_{sr}) \right]$$
(4)

$$P_r^{TCSC} = V_s^2 \Delta G_{sr} - V_s V_r \left[\Delta G_{sr} \cos(\delta_{sr}) - \Delta B_{sr} \sin(\delta_{sr}) \right]$$
(5)

$$Q_s^{TCSC} = -V_s^2 \Delta B_{sr} - V_s V_r \left[\Delta G_{sr} Sin(\delta_{sr}) - \Delta B_{sr} Cos(\delta_{sr}) \right]$$
(6)

$$Q_r^{TCSC} = -V_r^2 \Delta B_{sr} + V_s V_r [\Delta G_{sr} Sin(\delta_{sr}) + \Delta B_{sr} Cos(\delta_{sr})]$$
(7)

Where

$$\Delta G_{sr} = \frac{X_C R_{sr} (X_C - 2X_{sr})}{\left(R_{sr}^2 + X_{sr}^2\right) \left(R_{sr}^2 + (X_{sr} - X_C)^2\right)}$$
$$\Delta B_{sr} = \frac{-X_C \left(R_{sr}^2 - X_{sr}^2 + X_C X_{sr}\right)}{\left(R_{sr}^2 + X_{sr}^2\right) \left(R_{sr}^2 + (X_{sr} - X_C)^2\right)}$$

TCSC device is modeled with power injection model so far by using the TCSC control variable. It is possible to calculate the complex power injected S_s^{TCSC} and S_r^{TCSC} at bus-s and bus-r respectively.

$$S_s^{TCSC} = P_s^{TCSC} + j Q_s^{TCSC}$$
$$S_r^{TCSC} = P_r^{TCSC} + j Q_r^{TCSC}$$

i. Installation cost

In this paper, device life time is considered for 15 years during the analysis. The Installation Cost (IC) of

$$IC_{TCSC} = \frac{C_{TCSC} \times S_{TCSC} \times 1000}{n \times 8760} \ \$/h \tag{8}$$

Where, C_{TCSC} =[0.0015S²-0.713S+153.75] \$/KVAr, and S_{TCSC} =|Q2|-|Q1| MVAr. Here, C_{TCSC} and S_{TCSC} are the cost and operating range of TCSC, Q₁ and Q₂ are the reactive power flows in the line without and with TCSC; 'n' is the device life time in years (15 years).

ii. Optimal location

In order to select appropriate branch for placement of FACTS devices, contingency analysis and sensitivity index method is used. Sensitivity index is for rankina introduced the optimal location. Sensitiveness of each branch towards the change in power transfer capability, voltage profile with respect to change in line reactance, reactive power injection at various branches, buses are identified. The sensitivity indexes at different branches have been evaluated to select appropriate branch for device location by using load flow program suitable to transmission networks. These sensitivity indexes reflect how the load ability of branch change with respect to line reactance, how the voltage profile can be change with respect to reactive power injection at a particular bus. Sensitivity indices are evaluated for the base case and arrange in descending order. The branch which corresponds to highest sensitivity index is highest severe contingency.

The power flows of highest severe contingency are considered as base case. From these flows, the branch with least power flow can be considered as the best location for the TCSC device. Because the transmission line has inductive reactance where as the TCSC is a series controlled device which can provide a capacitive reactance so that the total reactance of the branch which leads to the increase of load ability of the line where the device was located. The objective is, to increase the power transfer capability of the transmission line. The power transfer capability of the transmission line depends on the line reactance as well as bus voltages. Hence the load ability of the line can be increased either by reducing line reactance or increasing voltage profile. In order to reduce the reactance of the transmission line TCSC can be used. The voltage profile can be improved by injecting the reactive power at a particular bus where the voltage is minimum. The optimal location of the device is that where it gives maximum benefit for optimal size of FACTS devices placed at selected location. The contingency analysis and severity index based ranking procedures are explained as follows:

a. Contingency analysis

Usually, outage of one of the transmission lines can be considered as the contingency condition. This happens due to many reasons; it may be because of maintenance of the transmission lines, environmental conditions, etc. Because of this situation, the transmission lines get overloaded. This contingency analysis analyzes the system security and gives future directions for the proper planning and designing. Because of this analysis, the preventive and corrective measures under contingency conditions can be predicted. Usually, the contingency condition may be due to a transmission line or a generator failure or a transformer failure. Out of which, transmission line failure plays an important role to assess the power system security.

b. Performance Index and Ranking

It is not required to consider each and every line outage to assess power system security. Because, for a particular line outage, there are few lines and buses do not have line flow and voltage variations beyond the limits. If there are any line overloads and bus voltage variations then the outage line is considered as critical line. The critical lines or credible contingencies are identified by contingency analysis and only these critical lines need to be taken to assess the power system security. A parameter which is known as performance index used for security analysis. It indicates how much a particular outage might affect the security of power system. In general the performance index (PI) or severity index (SI) is defined as

Severity Index =
$$\sum_{l=1}^{N_{line}} \left(\frac{S_l}{S_l^{\max}}\right)^{2m}$$
(9)

Where, S_l and S_l^{max} are the MVA flow in line 'l' and MVA rating of the line 'l' respectively. 'm' is an integer exponent taken as 1.

The line flows are obtained by using power flow solution method. In Eqn. (9), the performance index is defined only based on over loaded lines.

III. Opf Problem Formulation

A novel cost objective function is formulated by combining TCSC investment cost (ICTCSC) along with the conventional fuel cost function. Because of the combining ICTCSC with fuel cost function, the system and the FACTs controller control variables are readjusted to optimize the combined fuel cost function. This objective function is optimized while satisfying system operating, practical and the device limits. The modified fuel cost function with TCSC investment cost can be expressed as follows:

$$Minimize \quad FC = \sum_{i=1}^{N_G} C_i(P_{Gi}) + IC_{TCSC}$$
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Where, 'FC' is the total generation cost, 'C_i (P_{Gi})' is the fuel cost function of the ith unit, 'P_{Gi}' is the power generated by the ith unit and 'N_G' is the total number of generating units.

a) Quadratic Fuel cost function

The conventional quadratic fuel cost function can be expressed as follows:

$$C_i(P_{G_i}) = a_i P_i^2 + b_i P_i + c_i$$
(11)

Where, a_i , b_i , and c_i are the fuel-cost coefficients of the i^{th} unit.

b) Non-Convex Fuel cost function

The conventional quadratic fuel cost function with valve-point effects can be considered as nonconvex /non-smooth fuel cost function and this can be expressed as follows:

$$C_i(P_{Gi}) = a_i P_i^2 + b_i P_i + c_i + \left| e_i \times \sin\left(f_i \times \left(P_i^{\min} - P_i\right)\right)\right| \quad (12)$$

Where e_i and f_i are the fuel cost-coefficients of the ith unit reflecting valve-point loading effects.

The above problem is solved while satisfying equality, in-equality constraints explained in earlier chapters and the following device limits.

c) Constraints

The objective functions are subjected to the following equality, inequality and practical constraints.

i. Equality constraints

These constraints are usually load flow equations described as

$$P_{Gk} - P_{Dm} - \sum_{m=1}^{NB} |V_k| |V_m| |Y_{km}| \cos(\theta_{km} - \delta_k + \delta_m) = 0$$
$$Q_{Gk} - Q_{Dm} + \sum_{m=1}^{NB} |V_k| |V_m| |Y_{km}| \sin(\theta_{km} - \delta_k + \delta_m) = 0$$

where, ' $P_{Gk},~Q_{Gk}$ ' are the active and reactive power generations at k^{th} bus, ' $P_{Dm},~Q_{Dm}$ ' are the active and reactive power demands at m^{th} bus, 'NB' is number of buses, $|V_k|,~|V_m|$ are the voltage magnitudes at k^{th} and m^{th} buses, ' $\delta_k,~\delta_m$ ' are the phase angles of voltages at k^{th} and m^{th} buses, $|Y_{km}|,~\theta_{km}$ are the bus admittance magnitude and its angle between k^{th} and m^{th} buses.

ii. *In-equality constraints* Generator bus voltage limits

$$V_{G_i}^{\min} \le V_{G_i} \le V_{G_i}^{\max}, \quad \forall i \in N_G$$

Active Power Generation limits

$$P_{G_i}^{\min} \leq P_{G_i} \leq P_{G_i}^{\max}; \qquad \forall i \in N_G$$

Transformers tap setting limits

$$T_i^{\min} \le T_i \le T_i^{\max}; \qquad i = 1, 2, \dots, n_t$$

Capacitor reactive power generation limits

$$Q_{Sh_i}^{\min} \le Q_{Sh_i} \le Q_{Sh_i}^{\max}, \quad i = 1, 2, \dots, n_C$$

Transmission line flow limit

$$S_{li} \leq S_{li}^{\max}, \quad i = 1, 2, \dots, N_{line}$$

Reactive Power Generation limits

$$Q_{G_i}^{\min} \le Q_{G_i} \le Q_{G_i}^{\max}; \qquad \forall i \in N_G$$

Load bus voltage magnitude limits

$$V_i^{\min} \le V_i \le V_i^{\max} \qquad i = 1, 2, \dots, N_{load}$$

TCSC limits
$$-0.8X_{line} \le X_{TCSC} \le 0.2X_{line}$$

where ' n_t ' is the total number of taps, ' n_c ' is the total number of VAr sources, ' N_{load} ' is the total number load buses.

d) Practical constraints

The following practical constraints are considered to analyze the effect on the considered objectives.

i. Ramp-rate limits

The constraints of the ramp-rate limits, the operating limits of the generators are restricted to operate always between two adjacent periods forcibly. The ramp-rate constraints are

$$\max\left(P_{G_i}^{\min}, P_i^0 - DR_i\right) \le P_{G_i} \le \min\left(P_{G_i}^{\max}, P_i^0 + UR_i\right) \text{ where, } P_i^0$$

is i^{th} unit power generation at previous hour. DR_i and UR_i are the respective down and up ramp-rate limits of i^{th} unit.

ii. Prohibited Operating Zone (POZ) limits

To improve the efficiency of the thermal power plant generators are avoid to operate in the prohibited operating zones. This can be represented as

$$P_{i} = \begin{cases} P_{i}^{\min} \leq P_{i} \leq P_{i,1}^{L} \\ P_{i,k-1}^{U} \leq P_{i} \leq P_{i,k}^{L} \\ P_{i,n_{i}}^{U} \leq P_{i} \leq P_{i}^{\max} \end{cases}; k = 2,3,...,n_{i}$$

where n_i is the number of POZ's and k-index of POZ of unit-i. $P_{i,k}^L$ and $P_{i,k}^U$ are the respective lower and upper limit of kth POZ of ith generator.

Finally the above proposed problem is more generalized to solve in-equality constraints can be given as

$$FC_{aug} = FC + R_1 \left(P_{g,slack} - P_{g,slack}^{\lim} \right)^2 + R_2 \sum_{i=1}^{N_{Load}} \left(V_i - V_i^{\lim} \right)^2$$
$$+ R_3 \sum_{i=1}^{N_G} \left(Q_{G_i} - Q_{G_i}^{\lim} \right)^2 + R_4 \sum_{i=1}^{N_{Line}} \left(S_{l_i} - S_{l_i}^{\max} \right)^2$$

Where, R1, R2, R3 and R4 are the penalty quotients having large positive value. The limit values are defined as

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$$x^{\lim} = \begin{cases} x^{\max}, & x > x^{\max} \\ x^{\min}, & x < x^{\min} \end{cases}$$

Here ' x ' is the value of $\mathsf{P}_{g,\text{slack,}}\,\mathsf{V}_{\text{i}},\,\mathsf{Q}_{\text{Gi}}.$

IV. Imperialistic Competitive Algorithm

Imperialistic competitive algorithm [33] is inspired by the imperialistic competition in geo-political interactions between countries. Initially, countries for the considered control variables are generated. Out of which, some of them are best countries (lowest cost) treated as "imperialist" and the remaining are treated as "colonies". All colonies are moved towards their imperialists based on their powers. Here the power of each country is inversely proportional to its cost value. Finally, the "empires" are formulated by combining imperialists with the corresponding colonies.

After this, the assimilation policy is applied to move the empires towards their imperialist. Then power of each empire is calculated as the sum of the power of the imperialist and percentage of mean of power of its colonies. Then, all these empires participate in imperialistic competition and finally, the empire which has least power is eliminated from the system. The colonies will move towards their relevant imperialist and cause all the countries to converge to a state with single empire in the process.

The important steps in this algorithm are briefly discussed below:

a) Generating initial empires

Initially population is generated for all control variables as countries (Ncountry). For N-dimensional problem, the position of ith country is defined as follows:

Country_i =
$$[P_{G_1}, ..., P_{G_{NG}}, V_{G_1}, ..., V_{G_{NG}}, T_1, ..., T_{nt}, Q_{sh_1}, ..., Q_{sh_{nc}}]$$

The control variables corresponds to each population are updated in bus and line data then perform load flow and finally calculate the cost (C_i) of each country. Initialize the total number of imperialists (N_{imp}) and there by calculate the number of colonies $(N_{\text{Col}}=N_{\text{Country}}\text{-}N_{\text{imp}})$. To divide the colonies among imperialists proportionally, the normalized cost of all imperialists is calculated. From this the number of colonies for nth empire is evaluated. As the imperialists force to move the colonies towards them by applying attraction policy.

b) Moving colonies towards their imperialists

If a colony has best cost value than that of the imperialist, then exchange these colony and imperialist to continue this process in new location.

c) Calculation total power of an empire

The total power of an empire is the sum of the power of the imperialist and powers of the colonies.

d) Imperialistic competition

All these empires try to take the possession of colonies of other empires and try to control them. In this process, the power of the powerful empire increases where as the weak empire decreases. This competition is modeled by choosing some of the weakest colonies of the weakest empires and competition among all empires to possess these colonies. Then, total power of each empire is calculated.

e) Eliminating powerless empires

The powerless empires will collapse in the imperialistic competition. Different criteria can be defined for collapse mechanism. In this paper, an empire is assumed collapsed when it loses all of its colonies. Weak empires gradually decline in imperialistic competition and strong empires take the possession of their colonies. There are different conditions for declining an empire.

f) Stopping criteria

After some imperialistic competitions, all the empires except the most powerful one will collapse all of the countries under their possession become colonies of this empire. All the colonies have the same positions and the same costs and there is no difference between the colonies and their imperialist. In such a case, the algorithm stops.

V. Results and Analysis

An IEEE 14 bus system has been considered as first example to find the efficacy of the proposed method without and with TCSC. In 14 bus test system, bus 1 is slack bus, while buses 2, 3, 6 and 8 are generator buses and other buses are load buses. The input parameters of the proposed ICA method are given in Table.1.

As a preliminary computation the contingency analysis and ranking is performed on the test system for one branch outage at a time. The over loaded line flows and ranking during contingency analysis are tabulated in Tables 2 and 3 respectively. In this paper, Rank 1 contingency is considered to investigate the effectiveness of the proposed method without and with TCSC device to eliminate the over loaded lines.

As the TCSC is a series controlled FACTS device a line in which least power will flow is required for the best location of TCSC. The required best location for the TCSC is determined from the rank 1 contingency by using severity index. The SI values for the test system are given in Table3. From this table it can be seen that the line outage 4-5 is the most severe which is represented as rank 1 contingency. From rank 1 contingency result in order of priority of the line for the location of TCSC is 3-4 based on the power flows intransmission lines under rank 1 contingency.

Table 1 : Input parameters for test example

Parameters	Quantity
Number of initial countries	1000
Number of initial imperialists	8
Number of decades	200
Revolution rate	0.05
Assimilation coefficient	0.2
Zeta (ζ)	0.02
Damp ratio	0.99
Uniting threshold limit	0.02

Table 2 : Overloaded lines of IEEE-14 bus system during contingency analysis

S. No	Outage line	Over loaded lines	Line Flow (MVA)	Line limit (MVA)	PI
		1-2	179.4369	150	1.1962
		2-3	89.1447	85	1.0488
		2-4	89.2352	85	1.0498
1	4-5	5-6	62.1180	45	1.3804
		6-11	17.4159	14	1.244
		6-13	23.7676	22	1.0803
		10-11	13.4437	12	1.1203
		1-2	157.5373	150	1.0502
	7-9	5-6	60.7119	45	1.3492
2		6-11	17.6322	14	1.2594
		6-13	24.1568	22	1.098
		10-11	13.0879	12	1.0907

Table 3 : Contingency ranking of IEEE-14 bus system

S. No	Outage line	Severity Index	Rank
1	4-5	8.1198	1
2	7-9	5.8475	2

The results of optimized values for all control variables for minimization of quadratic and nonconvex fuel cost objective using proposed ICA method with considering effect of practical constraints for without and with TCSC are tabulated in Table.4. The convergence characteristics of the proposed algorithm for all the cases are shown in Figs 3 and 4.

From Table.4, it is observed that, with TCSC, the quadratic fuel cost is decreased by 0.6855 \$/h and the non-convex cost is decreased by 0.3291 \$/h when compared to without device. It is also observed that, the total real power generation is increased with TCSC and consequently the system power losses also increased.

The ramp-rate and POZ limits followed by the generators are tabulated in Table.5. From this table, it is observed that, while minimizing quadratic fuel cost objective the 2nd generator is following up ramprate and operative above the POZ upper limit whereas while minimizing non-convex fuel cist objective 2nd generator is following down ramp-rate and operating below the POZ lower limit.

From figures 3 and 4, it is observed that proposed method with TCSC starts the iterative process with good initial value and reaches final best value in less number of iterations when compared to without device.

Table 4 : OPF results of IEEE-14 bus system for convex and non-convex costs

S.No	Parameter		Quadratic cost (\$/h)		Non-convex cost (\$/h)	
			Without	With TCSC	Without	With TCSC
1	Real power generation (MW)	PG1	168.7799	173.1421	213.2493	213.563
		PG2	50.3254	47.0367	23.2421	25.5652
		PG3	21.1789	20	20	20
		PG6	18.1432	16.3694	7.7983	5.3562
		PG8	8.6787	10.6415	5	5
2	Generator voltages (p.u.)	VG1	1.0886	1.1	1.0883	1.1
		VG2	1.0565	0.9164	1.0599	0.9
		VG3	0.9419	1.0264	1.007	0.9912
		VG6	1.0505	0.9628	0.9836	0.9739
		VG8	1.015	1.0307	1.0507	0.9224
3	Transfor mer tap setting (p.u.)	TAP4-7	1.0508	1.0321	1.1	0.9631
		TAP4-9	0.9362	0.9962	0.9579	0.9692
		TAP5-6	1.0188	0.9226	1.0773	0.9598
4	Shunt compensa tor (MVAr)	QC9	19.4419	25.5866	9.3157	17.9661

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5	X _{TCSC} , p.u.	-	-0.0227	-	0.0201
6	Total generation (MW)	267.1061	267.1897	269.2897	269.4844
7	Total power loss (MW)	8.1061	8.1897	10.2897	10.4844
8	Total generation cost (\$/h)	715.6184	714.9329	824.7462	824.4171

Table 5 : Ramp rates and POZ limits followed by the generators of quadratic cost with TCSC for IEEE-14 bus system

Gen. No	Quadratic cost (\$/h)		Non-convex cost (\$/h)		
	Without	With TCSC	Without	With TCSC	
1	Up, 2	Up, 2	Up, 2	Up, 2	
2	Up, 2	Up, 2	Down, 1	Down, 1	
3	Down, 1	Down, 1	Down, 1	Down, 1	
6	-	-	-	-	
8	-	-	-	-	

1-Below POZ lower limit 3-Equal to POZ lower limit UP-following up-ramp rate







Fig 4 : Convergence characteristics of non convex cost with TCSC for IEEE-14 bus system

2-Above POZ upper limit 4-Equal to POZ upper limit Down-following down-ramp rate

VI. CONCLUSION

The ICA method has been successfully employed to solve the optimal power problem with generator constraints and valve point loading effect. Using this proposed method, the power system objectives such as quadratic and non-convex fuel cost objectives are optimized while satisfying system operating and practical constraints. The power injection modeling of TCSC has been presented with its incorporation procedure in conventional NR load flow. An optimal location identification methodology of TCSC has been identified to enhance the system security in terms of transmission line loadings. Obtained numerical results for IEEE-14 bus test system confirms that, the proposed methodology can handle any type of the objectives and can be applied to solve any size of the system.

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An Improvement Strategy on Direction Cosine Matrix based Attitude Estimation for Multi-Rotor Autopilot

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Abstract- In present days, autonomous multi-rotor copters are increasingly becoming popular due to its advantages in terms of multi-purpose functionalities, robustness, high dynamic response and more significantly inexpensive costs of development and usage. However, real-time attitude estimation is a key component which needs to have further development for better control and precise navigation to drive these unmanned aerial vehicles reliably. This paper presented an approach to improve the method of attitude estimation by Direction Cosine Matrix for multi-rotor copters that is based on low cost MEMS inertial sensors, a magnetometer and a commercial GPS receiver. In connection with the Direction Cosine Matrix attitude estimation scheme, a novel algorithm design for dealing with limitations on attitude sensing accuracy and reducing latency in real-time is proposed. The design is insensitive to noise or loss of GPS signals. The viability of the proposed design is demonstrated by an experimental scenario with real-time attitude information under different observations. From the experiment, some issues have been noticed and their reasons have been discussed.

Index Terms: attitude estimation, unmanned aerial vehicle, VTOL, RC aircraft, multi rotor copter, hover, MEMS, IMU, DCM, roll, pitch, yaw, euler angles, autopilot, gyro bias, gyro drift, accelerometer, magnetometer, calibration iron distortion, tilt compensation, normalization.

GJRE-F Classification : FOR Code: 090699



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An Improvement Strategy on Direction Cosine Matrix based Attitude Estimation for Multi-Rotor Autopilot

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Abstract- In present days, autonomous multi-rotor copters are increasingly becoming popular due to its advantages in terms of multi-purpose functionalities, robustness, high dynamic response and more significantly inexpensive costs of development and usage. However, real-time attitude estimation is a key component which needs to have further development for better control and precise navigation to drive these unmanned aerial vehicles reliably. This paper presented an approach to improve the method of attitude estimation by Direction Cosine Matrix for multi-rotor copters that is based on low cost MEMS inertial sensors, a magnetometer and a commercial GPS receiver. In connection with the Direction Cosine Matrix attitude estimation scheme, a novel algorithm design for dealing with limitations on attitude sensing accuracy and reducing latency in real-time is proposed. The design is insensitive to noise or loss of GPS signals. The viability of the proposed design is demonstrated by an experimental scenario with real-time attitude information under different observations. From the experiment, some issues have been noticed and their reasons have been discussed.

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I. INTRODUCTION

hese days unmanned flying robots or unmanned aerial vehicles (UAVs) are gaining popularity because of their wide range of applications. One type of UAV with a strong potential for both indoor and outdoor applications, is the multi-rotor copter. This special type of rotorcraft has the capability of vertical takeoff and landing (VTOL), as well as the ability to achieve rapid and stable motion in every direction. For these characteristics, multi-rotorcraft is suitable for hands-off autonomous operation within cluttered environments such as small buildings or caves including surveillance, and has many applications rescue, exploration in hazardous search and by human command, and therefore autopilot system with suitable control algorithm is required to make them stable. However, building a real-time attitude estimator in autopilot module for controlling multi-rotorcraft is the most critical challenge that engineers face while developing such aircraft. The attitude estimator is widely known as inertial measurement unit (IMU). Lowcost IMU systems are mainly based on micro-electromechanical systems (MEMS) technology. A typical IMU is equipped with 3-axis accelerometers, 3-axis rate gyroscopes and 3-axis magnetometers. Due to the limitations in sensing technologies and embedded processing, an IMU system cannot perform with zero error. Developing the control law with a better attitude estimating method can effectively improve the system performance and flight stability. Several attitude estimation methods have been developed in this research field, and direction cosine matrix (DCM) based attitude estimation is one of them. An improvement strategy on this estimation method is depicted in this paper.

The DCM is the matrix of rotation between a Cartesian coordinate frame, which is rigidly fixed in the aircraft body, and a reference Cartesian coordinate frame. This matrix is also known as rotation matrix (denoted by R) [1]. The DCM is a 3×3 matrix, whose nine elements are not independent. Like any rotation matrix, DCM is a proper orthogonal matrix, which implies six constraints connecting the nine elements. In spite of the redundancy created by the constraints, the DCM is a convenient rotation representation because the equations that describe the vector measurement model and the aircraft kinematical model are linear in DCM [1]. The DCM IMU estimation is a popular attitude estimation method that has been developed for model airplanes by William Premerlani and Paul Bizard [2]. According to this method a rate gyro is required to derive attitudes by intearatina the rigid body kinematic equations. accelerometer is required to provide gravity direction and GPS is essential to get the YAW reference.

This paper is organized as follows. Section III addresses the problem definition. Section IV describes the improvement strategy on DCM based attitude estimation for multi-rotor autopilot which includes coordinate system modification, step by step formation of rotation matrix for the modified coordinate systems, experimental analysis on using magnetometer-heading instead of GPS-heading, discussion on magnetometer error compensation and fusing reference vectors from

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GPS, and development of a new attitude estimation algorithm. The experimental results and analysis on the developed system are presented and discussed in Section V to validate and evaluate the system's usability.

II. Nomenclature

Some of the notations which have been used in this paper to understand the mechanical and mathematical relationships are presented in this section:

DCM: Direction Cosine Matrix

KF: Kalman Filter

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- EKF: Extended Kalman Filter
- QUEST: Quaternion Estimator
- GPS: Global Positioning System
- CG: Center of gravity
- FPU: Flight Processing Unit
- IMU: Inertial Measurement Unit
- PID: Proportional Integral Derivative
- YAW: Twist or oscillation about a vertical axis
- VTOL: Vertical takeoff and landing
- Attitude: Roll, pitch and yaw
- Heading: Aircraft's nose direction
- R: Rotation Matrix
- φ: Roll angle
- θ : Pitch angle
- ψ: Yaw angle
- $\boldsymbol{\omega}$: Angular Rate Vector
- W: Gyro sensor readings
- R_e : Earth frame, which is attached to the ground
- x_e, y_e, z_e : Axes of earth frame
- R_b : Body fixed frame, which is attached to the aircraft
- x_b, y_b, z_b : Axes of body fixed frame
- $(X, Y, Z)_{MAG}$: Measured values of X, Y, Z axes
- (X, Y, Z)_{COMP}: Compensated values of X, Y, Z axes
- ω_x : Angular rate on x-axis in R_b -frame
- a_x : Acceleration on x-axis in R_b -frame
- $I_3: 3\times 3$ Identity matrix

III. PROBLEM DEFINITION

In real world, the original technique of DCM based attitude estimation suffers some unavoidable problems. As any of these problems can cause occasional malfunctions, it cannot be used as a standalone process without support from a strong reference. The initial task of this process is to gather raw

roll, pitch and yaw. As gyroscope drifts over long time periods, it is necessary to take reference attitude data repeatedly to minimize its errors before calculating the final flight angles. So the accelerometer is used to get the reference vector of roll and pitch and the GPS is used to get the reference vector of yaw. GPS signal availability is essential for running the whole process because the IMU will lose its track over time without GPS. Since the transient response of GPS is not fast enough, the original technique of DCM based attitude estimation is not suitable for VTOL aircrafts. VTOL aircrafts are dynamically very sensitive along their yaw axis and this makes controlling the movement along yaw axis very hard. It is not possible to overcome this issue with any slow process like GPS. Magnetometer is a device for locating the direction of earth's magnetic poles with faster transient response than GPS system. So it can be used as the reference sensing element for yaw movement in VTOL aircraft. But in most cases, output data from magnetometer has offset values so using this device cannot help the estimation process with accurate data like GPS. Magnetometer outputs true values of earth's magnetic field only when it stays parallel to the ground and that means magnetometer will give wrong orientation data when the aircraft stays tilted to the ground. Also, magnetometer data gets distorted because of the nearby power lines as erroneous data will be produced if magnetometer cannot distinguish earth's true magnetic field due to interference of external

attitude data from gyroscope which includes angles of

In today's market, the coordinate system followed by the orientation sensing devices does not match with the conventional coordinate system used in the original DCM IMU estimation method and this will cause inconvenience for any autopilot system developer. Another drawback of this method is that it takes a lot of time to process the output.

DCM based attitude estimation method is more stable and provides accurate result than any other method. But still it is not suitable for multi-rotor crafts due to the issues mentioned above. So improving its methodology is necessary to use it for multi-rotor crafts.

IV. Improvement Strategy

The objective of improving the DCM based attitude estimation process by developing its algorithm to the next level of advancement is to make it suitable to use in real-time control processor of any autopilot module of multi-rotorcraft. The step-by-step development approaches are described below.

a) Coordinate System Modification

magnetic and electric fields.

The conventional coordinate system that was used to develop the rotation matrix of DCM IMU algorithm consists of two Cartesian coordinate frames, as shown in figure 1, one frame (x_b, y_b, z_b) is rigidly fixed

to the aircraft body, and the other one (x_e, y_e, z_e) is the earth's frame of reference.



Fig.1: The conventional coordinate system [2]

In conventional coordinate system the x_b and x_e axes, y_b and y_e axes, and z_b and z_e axes are directed towards forward, right, and downward direction respectively. If both the frames are rotated 180 degree along forward axis then the coordinate system will match with the coordinate system followed by the commercially available IMU hardware. So the inertial axes about the centre of gravity (CG) of aircraft body are:

- x_b axis positive forward
- y_b axis positive to left, perpendicular to x_b axis
- z_b axis positive upwards, perpendicular to x_b - y_b plane

According to the Euler theorem, the orientation of the aircraft body fixed frame (R_b) with respect to the earth fixed frame (R_e) can be described by three consecutive rotations of R_b along each of its axis, whose order is arbitrary, but the same axis may not be used twice in succession. The rotation sequences are usually denoted by three numbers, 1 for x-axis of R_b , 2 for y-axis of R_b , and 3 for z-axis of R_b . Among the twelve valid rotation sequences, we have proposed to choose the rotation sequence 132 or x-z-y for obtaining the attitude of aircraft in modified coordinate system. So the three consecutive rotations have been accomplished in the following order, assuming that the aircraft is initially positioned in such a way that the body frame is parallel to the earth frame:

- Rotating the body about x_b axis through roll angle (φ)
- Rotating the body about z_b axis through yaw angle $(-\psi)$
- Rotating the body about y_b axis through pitch angle (-θ)



Fig. 2: The modified coordinate system

b) Rotation Matrix Formation

In the previous section, the rotation sequence that has been proposed is the 132 or x-z-y. So considering a rotation from F reference frame to F^{'''} reference frame, the first rotation that is illustrated in Fig. 3, is about x-axis through an angle θ_x which is positive according to the right hand rule about the x-axis. With two rotations to go, the resulting alignment in general is oriented with neither of F or F^{'''}, but some intermediate reference frame (the first of two) denoted F'. Since the rotation was about x, x' is parallel to it but neither of the other two primed axes are.



Fig 3 : Anti-clockwise rotation from the reference frame F to F'

Now to describe this rotation, we can write:

$$\begin{bmatrix} F_{x'} \\ F_{y'} \\ F_{z'} \end{bmatrix} = \begin{bmatrix} \cos_{xx'} & \cos_{xy'} & \cos_{xz'} \\ \cos_{yx'} & \cos_{yy'} & \cos_{yz'} \\ \cos_{zx'} & \cos_{zy'} & \cos_{zz'} \end{bmatrix} \begin{bmatrix} F_{x} \\ F_{y} \\ F_{y} \\ F_{z} \end{bmatrix}$$
$$= \begin{bmatrix} \cos 0^{o} & \cos 90^{o} & \cos 90^{o} \\ \cos 90^{o} & \cos \theta_{x} & \cos(90^{o} + \theta_{x}) \\ \cos 90^{o} & \cos(90^{o} - \theta_{x}) & \cos \theta_{x} \end{bmatrix} \begin{bmatrix} F_{x} \\ F_{y} \\ F_{z} \end{bmatrix}$$
$$\therefore \begin{bmatrix} F_{x'} \\ F_{y'} \\ F_{z'} \end{bmatrix} = \begin{bmatrix} 1 & 0 & 0 \\ 0 & \cos \theta_{x} & -\sin \theta_{x} \\ 0 & \sin \theta_{x} & \cos \theta_{x} \end{bmatrix} \begin{bmatrix} F_{x} \\ F_{y} \\ F_{z} \end{bmatrix} = R_{x}(\theta_{x})$$
(1)

The next rotation that is shown in Fig. 4, is through an angle $-\theta_z$ about the z'-axis of the first intermediate reference frame to the second intermediate

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reference frame, F". Angle θ_z is negative according to the right hand rule about the z-axis. Note that $z^{"} = z'$, and neither x" or z" are necessarily axes of either F or F".



Fig. 4 : Clockwise rotation from the reference frame F' to F"

For this rotation, we can write:

$$\begin{bmatrix} F_{x''} \\ F_{y''} \\ F_{z''} \end{bmatrix} = \begin{bmatrix} \cos_{x'x''} & \cos_{x'y''} & \cos_{x'z''} \\ \cos_{y'x''} & \cos_{y'y''} & \cos_{y'z''} \\ \cos_{z'x''} & \cos_{z'y''} & \cos_{z'z''} \end{bmatrix} \begin{bmatrix} F_{x'} \\ F_{y'} \\ F_{z'} \end{bmatrix}$$
$$= \begin{bmatrix} \cos(-\theta_z) & \cos(90^\circ - \theta_z) & \cos 90^\circ \\ \cos(90^\circ + \theta_z) & \cos(-\theta_z) & \cos 90^\circ \\ \cos 90^\circ & \cos 90^\circ & \cos 90^\circ \end{bmatrix} \begin{bmatrix} F_{x'} \\ F_{y'} \\ F_{z'} \end{bmatrix}$$
$$\therefore \begin{bmatrix} F_{x''} \\ F_{y''} \\ F_{y''} \\ F_{z''} \end{bmatrix} = \begin{bmatrix} \cos \theta_z & \sin \theta_z & 0 \\ -\sin \theta_z & \cos \theta_z & 0 \\ 0 & 0 & 1 \end{bmatrix} \begin{bmatrix} F_{x'} \\ F_{y'} \\ F_{z'} \end{bmatrix} = R_y(\theta_y) \quad (2)$$

The final rotation that is illustrated in Fig. 5 is about y" through angle $-\theta_y$ and the final alignment is parallel to the axes of F". Angle θ_y is negative according to the right hand rule about the y-axis.



For describing the final rotation, we can write:

$$\begin{bmatrix} F_{x}^{"''} \\ F_{y}^{"''} \\ F_{z}^{"''} \end{bmatrix} = \begin{bmatrix} \cos_{x}^{"} x^{"''} & \cos_{x}^{"} y^{"''} & \cos_{y}^{"} z^{"'} \\ \cos_{y}^{"} x^{"'} & \cos_{y}^{"} y^{"'} & \cos_{y}^{"} z^{"'} \\ \cos_{z}^{"} x^{"'} & \cos_{z}^{"} y^{"''} & \cos_{z}^{"} z^{"'} \end{bmatrix} \begin{bmatrix} F_{x}^{"} \\ F_{y}^{"} \\ F_{z}^{"} \end{bmatrix}$$

$$\begin{bmatrix} \cos(-\theta_{y}) & \cos 90^{o} & \cos\{-(90^{o} - \theta_{y})\} \\ \cos 90^{o} & \cos 0^{o} & \cos 90^{o} \\ \cos(90^{o} - \theta_{y}) & \cos 90^{o} & \cos(-\theta_{y}) \end{bmatrix} \begin{bmatrix} F_{x}^{"} \\ F_{y}^{"} \\ F_{z}^{"} \end{bmatrix}$$

$$\therefore \begin{bmatrix} F_{x}^{"'} \\ F_{y}^{"'} \\ F_{z}^{"'} \end{bmatrix} = \begin{bmatrix} \cos \theta_{y} & 0 & -\sin \theta_{y} \\ \sin \theta_{y} & 0 & \cos \theta_{y} \end{bmatrix} \begin{bmatrix} F_{x}^{"} \\ F_{y}^{"} \\ F_{z}^{"} \end{bmatrix} = R_{z}(\theta_{z}) \quad (3)$$

It is known that the DCM is the product of three sequential rotations, so we can derive our desired DCM using the above results of all three sequential rotations. Later, the relation between the direction cosine matrix and rotation angles was found by replacing angle θ_x , θ_y , and θ_z by φ (roll angle), θ (pitch angle) and ψ (yaw angle) respectively. Finally the resultant matrix of direction cosines was used to rotate vectors, such as directions, velocities, accelerations, and translations by multiplying with each other. However, the R matrix is also called the DCM, because each entry is the cosine of the angle between an axis of the plane and an axis on the ground:

c) Magnetometer-Heading Experiment

A three-axis magnetometer has been used instead of using the GPS as the primary source to get the reference vector of yaw. In the process, the gyro sensor has been used to get the total orientation change along all three axes which include change in the yaw axis and the magnetometer checks the course over ground in every 125ms and detects the drifts of yaw information of gyro sensor for canceling out the drift to achieve a yawlock performance. After analyzing the timing for measuring input, we have found that magnetometer took 1.289ms for each measurement [3].

Fig 5 : Clockwise rotation from the reference frame F" to F"

$$R = R_X(\theta_X) \times R_Y(\theta_Y) \times R_Z(\theta_Z)$$

$$= \begin{bmatrix} \cos\theta_y \cos\theta_z & \sin\theta_z & -\sin\theta_y \cos\theta_z \\ -\cos\theta_y \cos\theta_x \sin\theta_z + \sin\theta_y \sin\theta_x & \cos\theta_x \cos\theta_z & \sin\theta_y \cos\theta_x \sin\theta_z - \cos\theta_y \sin\theta_x \\ \cos\theta_y \sin\theta_x \sin\theta_z + \sin\theta_y \cos\theta_x & \cos\theta_z \sin\theta_x & \sin\theta_y \sin\theta_x \sin\theta_z - \sin\theta_y \cos\theta_z \end{bmatrix}$$

$$\begin{bmatrix} \cos\theta \cos\psi & \sin\psi & -\sin\theta \cos\psi \\ \sin\theta \sin\psi + \sin\theta \sin\phi & \cos\phi \cos\psi & \sin\theta \cos\phi \sin\psi - \cos\theta \sin\phi \\ \cos\theta \sin\psi + \sin\theta \cos\phi & \cos\psi & \sin\theta \sin\phi \sin\psi - \sin\theta \cos\psi \end{bmatrix} = \begin{bmatrix} R_{xx} & R_{xy} & R_{xz} \\ R_{yx} & R_{yy} & R_{yz} \\ R_{zy} & R_{zy} & R_{zz} \end{bmatrix}$$
(4)

(6)



Fig. 6: Timing Analysis of HMC5843 Magnetometer [3]

To analyze the magnetometer data, we have collected real time sensor readings for each of the three axes over time under specific orientations such as positive x-axis, positive y-axis, and positive z-axis, all of them were separately towards North direction. Figure 7 shows the superimposition of measured magnetic fields for all three of the above mentioned arrangements:





b) Magnetometer Error Compensation

A low cost magnetometer works very well in clean magnetic environments, but this device is strongly influenced by its orientation or any magnetic perturbations produced by nearly available ferrous material or radio waves from mobile phone, loud speakers.

In the magnetometer-heading experiment it has been observed that the magnetometer needs to be held flat to function properly. Tilting would result in less accurate readings because magnetometer only uses two of the axes of earth's magnetic field. When this device is not parallel to these axes, the amount of magnetism felt by it will change based on the misalignment of the magnetometer to these axes. Magnetic disturbances were also observed during the experiment which developed a constant magnetic field component along each axis of the sensor outputs. To compensate these errors, the following processes were executed:

i. Tilt compensation

It is necessary to include the third axis, z, in magnetometer-heading calculation which can provide the magnetic field lost by the other two axes when they are out of alignment. So at first, a triple axis accelerometer has been used in parallel with the magnetometer to know how much the magnetometer is tilted. This was done for integrating the z-axis measurement properly to correct the tilt. The roll and angles which can be calculated pitch from accelerometer data, are never 0° when the device is tilted. If X_{MAG} , Y_{MAG} and Z_{MAG} are the measurement values for their respective axes (x, y, z) of the magnetometer and X_{COMP} and Y_{COMP} the are compensated values, then:

$$X_{COMP} = X_{MAG} . \cos(pitch) + Z_{MAG} . \sin(pitch)$$
(5)

$$Y_{COMP} = X_{MAG} \cdot sin(roll) \cdot sin(pitch) + Y_{MAG} \cdot cos(roll) - Z_{MAG} \cdot sin(roll) \cdot cos(pitch)$$

So, Heading = arctan
$$(Y_{COMP} / X_{COMP})$$
 (7)

i. Iron distortion compensation

Iron distortions can be categorized as two types: hard iron and soft iron effects. Hard iron distortions arise from permanent magnets, magnetized iron or steel nearby the magnetometer and remain constant and in a fixed location relative to the magnetometer for all heading orientations. To compensate for hard iron distortion, the offset in the origin of each of the reading scale is necessary to be determined. So for all three axes, these offsets were calculated by taking into account the minimum and maximum values as Eqn. 8 and using it on the readings of magnetometer on all three axes after rotating the magnetometer platform in a circle for each of the three axes [4]

$$OFFSET_{Reading} = \frac{MAX_{Reading} + MIN_{Reading}}{2}$$
(8)

The soft iron distortion is the effect of electromagnetic fields, which causes the ideal sphere of the 3D environment created by all the readings of the magnetometer to become an oval shape. It is necessary to perform the soft iron calibration in the firmware after the magnetometer values have been read and hard iron calibration has been done.

e) Fusing References from GPS

GPS provides a drift-free reference vector for the yaw orientation of the aircraft over the long term. The reasons why GPS has not been used as the primary source of reference vector for yaw information are:

- The transient response of the magnetometer is much faster than the GPS.
- The GPS device must move in order to provide direction information.

• The yaw reference cannot be updated under GPS denial environment.

In the attitude estimation process, magnetometer has been used for yaw reference vector and over a period of time the readings are jammed with offset values for various reasons. So when the GPS is available to calculate the heading information, the course over ground is collected from GPS and used to compensate the readings of magnetometer. While using the GPS signal, reporting latency and filtering have been taken into account. The procedure that has been followed to obtain the course over ground information using a low-cost GPS receiver is:

- *First:* Three consecutive GPS positions had been measured.
- *Second:* Those position measurements had been compared to obtain two velocity measurements
- *Third:* The velocity measurements had been used to calculate the average velocity over a short period.
- *Finally:* The course over ground had been calculated from average:

$$Heading = \arctan \left(Y_{velocity} / X_{velocity} \right)$$
(9)

f) Attitude Estimation Algorithm Design

After taking all these approaches into account, a new development over the attitude estimation algorithm of [5] have been proposed for better flight performance of multi-rotor aircrafts. The proposed attitude estimation algorithm calculates the orientation of an aircraft, in respect to the rotation of the earth by using rotation matrices. Figure 8 is a visual representation of the proposed attitude estimation algorithm. The input devices that were used in the algorithm are:

- Triple-axis gyro sensor: It measures angular velocity.
- Triple-axis accelerometer: It measures earth's gravitational field minus acceleration.
- Triple-axis magnetometer: It measures earth's magnetic field.
- GPS receiver: It measures drift free orientation and position information.

In this algorithm, the gyro sensor is the primary sensor used to calculate the orientation of a rigid body. Since gyro sensor's readings have different offsets depending on which direction the gyro sensor is facing; when these readings are integrated

over time, it causes the integral result to drift. The accelerometer is not affected by drift; therefore, it has been used as an orientation reference of that rigid body to compensate the roll-pitch error (i.e. gyro's offset error). After error compensation, the magnetometer's readings has been used to calculate the heading (i.e. yaw movement) of that rigid body. To refresh the measured yaw information with true values over long time period, the GPS signal has also been used when the signal is available and ready to generate valid yaw information.



Fig. 8: Proposed attitude estimation algorithm for multi rotor aircrafts

Therefore, accelerometer, magnetometer and GPS receiver all these devices allow the system to calculate the rotation correction matrix.

Afterwards, a proportional plus integral feedback controller has been used on the correction matrix to remove the drift from the gyro sensor's readings. The compensated gyro sensor's readings, W, are then passed onto another process known as normalization and kinematics. The rotation matrix's columns are unit vectors which is why the gyro readings have been normalized before calculating the kinematics portion. Once normalized, the gyro readings along with the previous rotation matrix are used to calculate the current rotation matrix by using Equation 17 for updating the direction cosine matrix from gyro signals in the 'Computing Direction Cosines From Gyro Signals' section in [1]. Finally, the rotation angles (roll, pitch and yaw) are calculated from the updated rotation matrix by the following equations which were generated from Equation 4:

$$\varphi = a \tan 2 \left(R_{zy}, R_{yy} \right) \tag{10}$$

$$\theta = - \operatorname{a} \tan 2 \left(R_{xz}, R_{xx} \right) \tag{11}$$

$$\psi = \arcsin \left(R_{xy} \right) \tag{12}$$

V. EXPERIMENTAL RESULTS AND ANALYSIS

The proposed design of attitude estimation algorithm is required to validate and evaluate before considering its usability in real world. This is why realtime attitude reading test has been performed under different conditions. A 9DoF MEMS based IMU hardware setup has been made by following the modified coordinate system. This setup includes a 3-axes digital gyro sensor - ITG3200, a 3-axes digital accelerometer -BMA180 and a 3-axes magnetometer - HMC5843. Along with a 5Hz 66-channel GPS receiver -LS20031, the 9DoF IMU setup has been integrated into a guad-rotor copter that can hover in the air for more than 10 minutes. In this test, the real-time attitude data from the implemented prototype have been passed through the wireless serial data communication with 115200 baud rate to computer for performance observation in Matlab.



Fig. 9 : Real-time Attitude Readings

The procedure of the test is, first the system remained static for calculating the real-time attitude initials on ground. During this part of the test, no calibration was done so that the impact of the algorithm on unprocessed input data from sensor with random offset values can be observed. By writing a script, 5700 samples of the measured data has been collected in Matlab for 10 minutes and plotted on a graph (Fig. 9) with a time vector.



After the attitude calculation converges with the aid of proper calibration process, the results found were more precise, static and valid for the same event. Figure 10 show that the estimated roll and pitch angle stays zero after some random fluctuations for a few seconds. Meanwhile, the yaw angle also stays static with a very little value.

In the final part of the test, the device was flown and allowed to hover throughout the test. Meanwhile, the readings were captured in Matlab simultaneously. The duration of this test was also 10 minutes (600 seconds). The real-time attitude estimation depicted in Fig. 11, portrays that estimated attitude information holds a very little noise which occurred mostly due to the disturbance from external sources. It was observed that the roll and pitch angle remained much more stable than the yaw parameter and the yaw reading behaved like a wave.



Fig. 11: Real-time Attitude Readings at hover

One of the reasons which caused this wave is the slight time delay between readings of MEMS sensors and the reference from GPS. I

rotation rate errors consist of the constant bias and the random noise. During the test, GPS observation is available every 200ms. Therefore the attitude data derived from MEMS sensors updated by GPS are at a rate of 5Hz.

From these test results, it can be concluded that the proposed algorithm can impressively deal with the MEMS gyro drifts with better consistency. So, this test validates the proposed design algorithm and strategy.

VI. CONCLUSIONS

This paper presented an improved attitude estimation method based on the Direction Cosine Matrix. This estimator is built for attitude control and stabilization of multi-rotor autopilot systems. The unresolved issues that still exist can be corrected and considered for future improvements. Future applications will be presented with the integration of position, attitude and altitude estimation based on GPS/IMU/computer vision/ ultrasonic sound, resulting in an estimator of twelve degrees of freedom that will be used for tracking and navigation.

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The Peak to Average Power Reduction (PAPR) Technique of OFDM Signal by using Clipping and Filtering Method

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Abstract- Orthogonal Frequency Division Multiplexing (OFDM) has several attributes which make it a preferred modulation scheme for high speed wireless communication. Orthogonal Frequency Division Multiplexing (OFDM) has been currently under intense research for broadband wireless transmission due to its robustness against multipath fading. However OFDM signals have a problem with high Peak-to- Average power ratio (PAPR) and thus, a power amplifier must be carefully manufactured to have a linear input-output characteristic or to have a large input power back-off. In recent years, many peak-to-average power ratio (PAPR) reduction techniques have been proposed for orthogonal frequency division multiplexing (OFDM) signals. Among various techniques, the clipping and filtering technique has been considered as a practical scheme, and widely used owing to its non-expansion of bandwidth, low computational complexity, and simplicity in implementation without receiver-side cooperation.

Keywords: CCDF, CDF, CDMA, DAB, DVB, FFT, IFFT, ISI, LAN, MCM, OFDM, PAPR.

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The Peak to Average Power Reduction (PAPR) Technique of OFDM Signal by using Clipping and Filtering Method

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Keywords: CCDF, CDF, CDMA, DAB, DVB, FFT, IFFT, ISI, LAN, MCM, OFDM, PAPR.

I. INTRODUCTION

division rthogonal frequency multiplexing (OFDM) is becoming the chosen modulation technique for wireless communications. Orthogonal Frequency Division Multiplexing (OFDM) can be termed as an alternative wireless modulation technology to CDMA. OFDM (Orthogonal Frequency Division Multiplexing) is a multicarrier modulation that is implemented in many recent wireless applications due to its ability to combat impulsive noise and multipath effects and make better use of the system available bandwidth. It has been adopted for the European Digital Audio Broadcasting (DAB) [1] and Digital Video Terrestrial Broadcasting (DVB) standards, it has been proposed for UMTS (Universal Mobile Telecommunication Systems) [2] and it has just been standardized for new wireless LAN generations (HIPERLAN: High Performance Radio LAN. OFDM offer high spectral efficiency, immune to the multipath delay, low inter-symbol interference (ISI), immunity to frequency selective fading and high power efficiency. Due to these merits OFDM is chosen as high data rate communication systems such as Digital Video Broadcasting (DVB) and based mobile worldwide interoperability for microwave access (mobile Wi-MAX)[3]. The basic principle of OFDM is to split a highrate data stream into a number of lower rate streams that are transmitted simultaneously over a number of subcarriers. These subcarriers are overlapped with each other. Because the symbol duration increases for lower rate parallel subcarriers, the relative amount of dispersion in time caused by multipath delay spread is decreased. Inter-symbol interference (ISI) is eliminated almost completely by introducing a guard time in every OFDM symbol [4]. The entire data stream of OFDM is divided into different blocks of N symbols each. Each block is multiplied with U different phase factors to generate U modified blocks before giving to IFFT block. Each modified block is given to different IFFT block to generate OFDM symbols. PAPR is calculated for each modified block and select the block which is having minimum PAPR ratio.

In this paper we have investigate OFDM signals, Clipping and Filtering method, OFDM signals with and without Clipping and Filtering method and compare between them.

II. Ofdm System Model

OFDM is a special form of multicarrier modulation (MCM) with densely spaced subcarriers with overlapping spectra, thus allowing multiple-access [5]. MCM works on the principle of transmitting data by dividing the stream into several bit streams, each of which has a much lower bit rate and by using these substreams to modulate several carriers. In multicarrier transmission, bandwidth divided in many nonoverlapping subcarriers but not necessary that all subcarriers are orthogonal to each other [5]. In OFDM the sub-channels overlap each other which leads to an efficient use of the total bandwidth. The information and sent over the N sub-channels, one symbol per channel. To permit dense packing and still ensure that a

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minimum of interference between the sub-channels is encountered, the carrier frequencies must be chosen carefully. By using orthogonal carriers, frequency domain can be viewed so as the frequency space between two sub-carriers is given by the distance to the first spectral null [6].





By converting a single high frequency carrier to several sub-carriers, OFDM enhances the ability to cope with frequency selective fading effects and narrow bandwidth interference. The orthogonal property also greatly simplifies the design of both transmitter and receiver. A receiver can detect every sub-carrier data, which commonly is done via Fast Fourier Transform (FFT). Therefore a separate filter for each sub channel is not required. However, in practice, the sub-carriers are modulated in different amplitude and phase [7].

a) Peak to Average Power Ratio(PAPR) for OFDM signal

The peak to average power ratio for a signal $\boldsymbol{x}(t)$ is defined as

$$\mathsf{PAPR} = \frac{\max [!x(t)x*(t)]}{\mathbb{E}[!x(t)x*(t)]} \tag{1}$$

Where ()^{*} corresponds to the conjugate operator [8]. Expressing in deciBels,

$$PAPR_{dB} = 10log_{10}(PAPR)$$
(2)

b) PAPR of a complex sinusoidal signal

Consider a sinusoidal signal $x(t) = e^{2\pi f t}$ having the period T. The peak value of the signal is

$$\max[x(t)x * (t)] = +1$$
 (4)

The mean square value of the signal is

$$E[x(t)x * (t)] = \frac{1}{T} \int_0^T exp^{4\pi ft} = 1$$
 (5)

Given so, the PAPR of a single complex sinusoidal tone is, PAPR=1[8].



c) Clipping and Filtering Technique

In OFDM, signal contains high peaks (exceeding a certain threshold) will be applied to clipping and Filtering processes (CAF) as illustrated in fig.-3. In the Clipping part, when amplitude exceeds a certain threshold, the amplitude is hard-clipped while the phase is saved [9-10].



Fig 3: Repeating clipping and filtering technique

In fig-3, vector $A_1 = [A_0, \dots, A_{N-i}]$, obtained after oversampling stage is first transformed using an oversize inverse fast Fourier transformation (IFFT). For an oversampling factor denoted by IF, A₁ is extended by adding N(IF-I) zeros in the middle of the vector. This results in a trigonometric interpolation of the signal time domain signal [I]. The interpolated signal is then clipped. In this paper, hard-limiting is applied to the amplitude of the complex values at the IFFT output. However, any other form of nonlinearity could be used. The clipping ratio, CR, is defined as the ratio of the clipping level value to the root mean square value of the unclipped signal. The clipping is followed by filtering to reduce out-of band power. The filter consists of two FFT operations. The forward FFT transforms the clipped signal back into the discrete frequency domain resulting in vector. The in- band discrete frequency components of, $[C_{0,j}, \dots, C_{N/2-1,i}, C_{N/1-N/2,j}, \dots, C_{N/1-1,i}]$, are passed unchanged to the inputs of the second IFFT while the out-of-band components ,[$C_{N/2+1,i,}...$ $C_{NI1\text{-}N/2,i}$] are nulled. In systems where some band-edge subcarriers are unused the components corresponding to these are also nulled. The resulting filter is a time-dependent filter, which passes in-band and rejects out-of-band discretefrequency components. This means that it causes no distortion to the in-band OFDM signal. Since the filter operates on a symbol-by-symbol basis, it causes no Inter-symbol interference. The filtering does cause some peak to re-growth. Clipping method sets a clipping threshold, when the amplitude of the signals over the threshold, then cut the high peak power. According to the system acquirement, the following function has been used to calculate the clipping ratio. PAPR0=10logCR, where, PAPRO is the threshold value, and CR is the clipping ratio. Due to the relation between PAPR0 and the system BER, PAPRO is selected to be inverse ratio to BER. In this case, proper threshold value should be selected carefully.

III. SIMULATION AND RESULTS

The Cumulative Distribution Function (CDF) is one of the most regularly used parameters, which is used to measure the efficiency of any PAPR technique. Normally, the Complementary CDF (CCDF) is used instead of CDF, which helps us to measure the probability that the PAPR of a certain data block exceeds the given threshold. The CCDF of the PAPR of the data block is desired in our case to compare outputs of various reduction techniques [7-8].

The simulation result of amplitude clipping method is shown in Fig-4. It can be observe that OFDM signal is has higher PAPR and after applying this method PAPR is reduced significantly. This PAPR is decreases as the number of clip and filtering is increased from one to two levels. Because the clipping is followed by filtering to reduce out of band power.



Fig 4 : Comparison of CCDF using one/two clipping and filtering method

IV. CONCLUTION

In conclusion, OFDM technology summed up a number of sub carriers modulated by group of data symbol. Therefore, transmitted signal may have a relatively large peak power which leads to high PAPR. The principal drawback of OFDM is that the peak transmitted power can be substantially larger than the average power. We observe that the PAR-reduction problem for OFDM has received a great deal of attention recently. In this paper, It can be observe that OFDM signal is has higher PAPR and after applying this method PAPR is reduced significantly. This PAPR is decreases as the number of clip and filtering is increased from one to two levels. Because the clipping is followed by filtering to reduce out of band power. The DFT transform the clipped signal into frequency domain signal.

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Language: The language of publication is UK English. Authors, for whom English is a second language, must have their manuscript efficiently edited by an English-speaking person before submission to make sure that, the English is of high excellence. It is preferable, that manuscripts should be professionally edited.

Standard Usage, Abbreviations, and Units: Spelling and hyphenation should be conventional to The Concise Oxford English Dictionary. Statistics and measurements should at all times be given in figures, e.g. 16 min, except for when the number begins a sentence. When the number does not refer to a unit of measurement it should be spelt in full unless, it is 160 or greater.

Abbreviations supposed to be used carefully. The abbreviated name or expression is supposed to be cited in full at first usage, followed by the conventional abbreviation in parentheses.

Metric SI units are supposed to generally be used excluding where they conflict with current practice or are confusing. For illustration, 1.4 I rather than $1.4 \times 10-3$ m3, or 4 mm somewhat than $4 \times 10-3$ m. Chemical formula and solutions must identify the form used, e.g. anhydrous or hydrated, and the concentration must be in clearly defined units. Common species names should be followed by underlines at the first mention. For following use the generic name should be constricted to a single letter, if it is clear.

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All manuscripts submitted to Global Journals Inc. (US), ought to include:

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Abstract, used in Original Papers and Reviews:

Optimizing Abstract for Search Engines

Many researchers searching for information online will use search engines such as Google, Yahoo or similar. By optimizing your paper for search engines, you will amplify the chance of someone finding it. This in turn will make it more likely to be viewed and/or cited in a further work. Global Journals Inc. (US) have compiled these guidelines to facilitate you to maximize the web-friendliness of the most public part of your paper.

Key Words

A major linchpin in research work for the writing research paper is the keyword search, which one will employ to find both library and Internet resources.

One must be persistent and creative in using keywords. An effective keyword search requires a strategy and planning a list of possible keywords and phrases to try.

Search engines for most searches, use Boolean searching, which is somewhat different from Internet searches. The Boolean search uses "operators," words (and, or, not, and near) that enable you to expand or narrow your affords. Tips for research paper while preparing research paper are very helpful guideline of research paper.

Choice of key words is first tool of tips to write research paper. Research paper writing is an art.A few tips for deciding as strategically as possible about keyword search:



- One should start brainstorming lists of possible keywords before even begin searching. Think about the most important concepts related to research work. Ask, "What words would a source have to include to be truly valuable in research paper?" Then consider synonyms for the important words.
- It may take the discovery of only one relevant paper to let steer in the right keyword direction because in most databases, the keywords under which a research paper is abstracted are listed with the paper.
- One should avoid outdated words.

Keywords are the key that opens a door to research work sources. Keyword searching is an art in which researcher's skills are bound to improve with experience and time.

Numerical Methods: Numerical methods used should be clear and, where appropriate, supported by references.

Acknowledgements: Please make these as concise as possible.

References

References follow the Harvard scheme of referencing. References in the text should cite the authors' names followed by the time of their publication, unless there are three or more authors when simply the first author's name is quoted followed by et al. unpublished work has to only be cited where necessary, and only in the text. Copies of references in press in other journals have to be supplied with submitted typescripts. It is necessary that all citations and references be carefully checked before submission, as mistakes or omissions will cause delays.

References to information on the World Wide Web can be given, but only if the information is available without charge to readers on an official site. Wikipedia and Similar websites are not allowed where anyone can change the information. Authors will be asked to make available electronic copies of the cited information for inclusion on the Global Journals Inc. (US) homepage at the judgment of the Editorial Board.

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Tables: Tables should be few in number, cautiously designed, uncrowned, and include only essential data. Each must have an Arabic number, e.g. Table 4, a self-explanatory caption and be on a separate sheet. Vertical lines should not be used.

Figures: Figures are supposed to be submitted as separate files. Always take in a citation in the text for each figure using Arabic numbers, e.g. Fig. 4. Artwork must be submitted online in electronic form by e-mailing them.

Preparation of Electronic Figures for Publication

Even though low quality images are sufficient for review purposes, print publication requires high quality images to prevent the final product being blurred or fuzzy. Submit (or e-mail) EPS (line art) or TIFF (halftone/photographs) files only. MS PowerPoint and Word Graphics are unsuitable for printed pictures. Do not use pixel-oriented software. Scans (TIFF only) should have a resolution of at least 350 dpi (halftone) or 700 to 1100 dpi (line drawings) in relation to the imitation size. Please give the data for figures in black and white or submit a Color Work Agreement Form. EPS files must be saved with fonts embedded (and with a TIFF preview, if possible).

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1. Choosing the topic: In most cases, the topic is searched by the interest of author but it can be also suggested by the guides. You can have several topics and then you can judge that in which topic or subject you are finding yourself most comfortable. This can be done by asking several questions to yourself, like Will I be able to carry our search in this area? Will I find all necessary recourses to accomplish the search? Will I be able to find all information in this field area? If the answer of these types of questions will be "Yes" then you can choose that topic. In most of the cases, you may have to conduct the surveys and have to visit several places because this field is related to Computer Science and Information Technology. Also, you may have to do a lot of work to find all rise and falls regarding the various data of that subject. Sometimes, detailed information plays a vital role, instead of short information.

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21. Arrangement of information: Each section of the main body should start with an opening sentence and there should be a changeover at the end of the section. Give only valid and powerful arguments to your topic. You may also maintain your arguments with records.

22. Never start in last minute: Always start at right time and give enough time to research work. Leaving everything to the last minute will degrade your paper and spoil your work.

23. Multitasking in research is not good: Doing several things at the same time proves bad habit in case of research activity. Research is an area, where everything has a particular time slot. Divide your research work in parts and do particular part in particular time slot.

24. Never copy others' work: Never copy others' work and give it your name because if evaluator has seen it anywhere you will be in trouble.

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26. Go for seminars: Attend seminars if the topic is relevant to your research area. Utilize all your resources.

27. Refresh your mind after intervals: Try to give rest to your mind by listening to soft music or by sleeping in intervals. This will also improve your memory.

28. Make colleagues: Always try to make colleagues. No matter how sharper or intelligent you are, if you make colleagues you can have several ideas, which will be helpful for your research.

29. Think technically: Always think technically. If anything happens, then search its reasons, its benefits, and demerits.

30. Think and then print: When you will go to print your paper, notice that tables are not be split, headings are not detached from their descriptions, and page sequence is maintained.

31. Adding unnecessary information: Do not add unnecessary information, like, I have used MS Excel to draw graph. Do not add irrelevant and inappropriate material. These all will create superfluous. Foreign terminology and phrases are not apropos. One should NEVER take a broad view. Analogy in script is like feathers on a snake. Not at all use a large word when a very small one would be sufficient. Use words properly, regardless of how others use them. Remove quotations. Puns are for kids, not grunt readers. Amplification is a billion times of inferior quality than sarcasm.

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33. Report concluded results: Use concluded results. From raw data, filter the results and then conclude your studies based on measurements and observations taken. Significant figures and appropriate number of decimal places should be used. Parenthetical remarks are prohibitive. Proofread carefully at final stage. In the end give outline to your arguments. Spot out perspectives of further study of this subject. Justify your conclusion by at the bottom of them with sufficient justifications and examples.

34. After conclusion: Once you have concluded your research, the next most important step is to present your findings. Presentation is extremely important as it is the definite medium though which your research is going to be in print to the rest of the crowd. Care should be taken to categorize your thoughts well and present them in a logical and neat manner. A good quality research paper format is essential because it serves to highlight your research paper and bring to light all necessary aspects in your research.

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Key points to remember:

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- Please note the criterion for grading the final paper by peer-reviewers.

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The introduction will be compiled from reference matter and will reflect the design processes or outline of basis that direct you to make study. As you will carry out the process of study, the method and process section will be constructed as like that. The result segment will show related statistics in nearly sequential order and will direct the reviewers next to the similar intellectual paths throughout the data that you took to carry out your study. The discussion section will provide understanding of the data and projections as to the implication of the results. The use of good quality references all through the paper will give the effort trustworthiness by representing an alertness of prior workings.

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In every sections of your document

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- \cdot Align the primary line of each section
- · Present your points in sound order
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The summary should be two hundred words or less. It should briefly and clearly explain the key findings reported in the manuscript-must have precise statistics. It should not have abnormal acronyms or abbreviations. It should be logical in itself. Shun citing references at this point.

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- Fundamental goal
- To the point depiction of the research
- Consequences, including <u>definite statistics</u> if the consequences are quantitative in nature, account quantitative data; results of any numerical analysis should be reported
- Significant conclusions or questions that track from the research(es)

Approach:

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- Center on shortening results bound background information to a verdict or two, if completely necessary
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- Present a justification. Status your particular theory (es) or aim(s), and describe the logic that led you to choose them.
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Approach:

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- Materials may be reported in a part section or else they may be recognized along with your measures.

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The page length of this segment is set by the sum and types of data to be reported. Carry on to be to the point, by means of statistics and tables, if suitable, to present consequences most efficiently. You must obviously differentiate material that would usually be incorporated in a study editorial from any unprocessed data or additional appendix matter that would not be available. In fact, such matter should not be submitted at all except requested by the instructor.



Content

- Sum up your conclusion in text and demonstrate them, if suitable, with figures and tables.
- In manuscript, explain each of your consequences, point the reader to remarks that are most appropriate.
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- Give details all of your remarks as much as possible, focus on mechanisms.
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- One research will not counter an overall question, so maintain the large picture in mind, where do you go next? The best studies unlock new avenues of study. What questions remain?
- Recommendations for detailed papers will offer supplementary suggestions.

Approach:

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Topics	Grades		
	А-В	C-D	E-F
Abstract	Clear and concise with appropriate content, Correct format. 200 words or below	Unclear summary and no specific data, Incorrect form Above 200 words	No specific data with ambiguous information Above 250 words
Introduction	Containing all background details with clear goal and appropriate details, flow specification, no grammar and spelling mistake, well organized sentence and paragraph, reference cited	Unclear and confusing data, appropriate format, grammar and spelling errors with unorganized matter	Out of place depth and content, hazy format
Methods and Procedures	Clear and to the point with well arranged paragraph, precision and accuracy of facts and figures, well organized subheads	Difficult to comprehend with embarrassed text, too much explanation but completed	Incorrect and unorganized structure with hazy meaning
Result	Well organized, Clear and specific, Correct units with precision, correct data, well structuring of paragraph, no grammar and spelling mistake	Complete and embarrassed text, difficult to comprehend	Irregular format with wrong facts and figures
Discussion	Well organized, meaningful specification, sound conclusion, logical and concise explanation, highly structured paragraph reference cited	Wordy, unclear conclusion, spurious	Conclusion is not cited, unorganized, difficult to comprehend
References	Complete and correct format, well organized	Beside the point, Incomplete	Wrong format and structuring

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