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Luis Galárraga

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Presentation of Strong Candidates for Dark Matter

By Koshun Suto

Chudaiji Buddhist Temple, Japan

Abstract- This paper presents atoms and molecules incorporating hydrogen at ultra-low energy levels as a strong candidate for Dark Matter. The existence of electrons at these energy levels can be demonstrated by changing the interpretation of triplet production. The radius of this undiscovered hydrogen atom is extremely small. The radius is about 1.331×10^{-5} the radius of an ordinary hydrogen atom in the 1s state. If many of these atoms or molecules are collected together, a state with extremely high density will be realized. This paper predicts that, in addition to such hydrogen, diverse types of atoms and various types of molecules comprised of diverse types of atoms, can also be candidates for Dark Matter.

Keywords: dark matter, dark hydrogen, ultra-low energy levels, triplet production, einstein's energy-momentum relationship, proton radius.

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I. INTRODUCTION

It is well-known that the existence of Dark Matter (DM) was first pointed out by F. Zwicky in 1933.¹ Zwicky was investigating the dispersion velocity of galaxies in the Coma Cluster, but that velocity was so large that it could not be explained with classical mechanics. Therefore, Zwicky posited the existence of an unknown form of matter to explain the reason for that velocity.

Furthermore, in the latter half of the 1970s, it was discovered that DM also exists in galaxies, and since then it has been revealed by a variety of observations. As a result, the existence of DM is supported by many scientists today. This unknown form of matter is thought to have the following characteristics:

- (1) It is widely present in galactic systems.
- (2) It is electrically neutral.
- (3) It has considerable mass.
- (4) It cannot be observed optically (it does not emit light).
- (5) It exhibits almost no interaction with matter.
- (6) It moves at a speed far slower than light.

DM candidates can be roughly divided into two types: elementary particle candidates and astrophysical candidates. The leading elementary particle candidate is a Weakly Interacting Massive Particle (WIMP).

The Kavli Institute for the Physics and Mathematics of the Universe (Kavli IPMU) has investigated the distribution of DM using observation data for 50 galactic clusters photographed with the

Subaru Telescope. As a result, they announced that the results agreed with a theory called Cold Dark Matter (CDM).²

In addition, a new theory was announced this year which posits another force dominating the world of the extremely small, and the existence of a Strongly Interacting Massive Particle (SIMP).³

At present, the CDM scenario is thought to have the advantage, but the fact that the candidate particle has not been discovered is regarded as a weak point.

Matter with the six characteristics given above does not exist among the elementary particles we currently know of.

The simplest model assumes one type of particle is involved in DM. However, since the 1990s various experiments have attempted to directly detect WIMPs, but no definitive signs suggesting the existence of WIMPs have been found. Therefore, some scientists have doubts about the current theory, and have begun to also consider models of DM comprised of multiple particles.^{4,5}

This paper believes that DM does not necessary have to be elementary particle. Last year, the author published a paper predicting the existence of hydrogen atoms at ultra-low energy levels.⁶ This paper examines whether such hydrogen atoms can be candidates for DM.

Prior to that, Sec. II explains the equations which are the departure point when deriving the equation for the ultra-low energy levels of the hydrogen atom. Sec. III actually derives the unknown energy levels from those equations. Sec. IV presents the experimental grounding demonstrating the existence of those energy levels.

II. EQUATION SERVING AS THE BASIS FOR DERIVING ULTRA-LOW ENERGY LEVELS OF THE HYDROGEN ATOM

One of the most important relationships in the special theory of relativity is as follows:

$$(mc^2)^2 = p^2c^2 + (m_0c^2)^2. \quad (1)$$

Here, m_0c^2 is the rest mass energy of an object or a particle, and mc^2 is the relativistic energy. Eq. (1) can also be applied to the electron, but in that case it becomes as follows:

$$(mc^2)^2 = p^2c^2 + (m_e c^2)^2. \quad (2)$$

Here, $m_e c^2$ is the rest mass energy of an electron.

However, since Eq. (2) does not include the potential energy of the electron, this equation cannot be applied to the electron in the hydrogen atom.

Incidentally, in the classical quantum theory of Bohr, the energy eigenvalues of the hydrogen atom can be expressed with the following equation:

$$E_{B,n} = -\frac{1}{2} \left(\frac{1}{4\pi\epsilon_0} \right)^2 \frac{m_e e^4}{\hbar^2} \frac{1}{n^2} = -\frac{\alpha^2 m_e c^2}{2n^2}, \quad n=1,2,\dots \quad (3)$$

Here, the B in E_B indicates the equation derived by Bohr. In this case, the total mechanical energy of the electron has a negative value.

However, according to quantum mechanics, the energy eigenvalues of a hydrogen atom as obtained from the Dirac relativistic wave equation are as follows.⁷

$$E = m_e c^2 \left[1 - \frac{\gamma^2}{2n^2} - \frac{\gamma^4}{2n^4} \left(\frac{n}{|k|} - \frac{3}{4} \right) \right]. \quad (4)$$

γ matches with the fine structure constant α . That is:

$$\alpha = \frac{e^2}{4\pi\epsilon_0 \hbar c} = 7.2974 \times 10^{-3}. \quad (5)$$

If we ignore for the third term of this equation, Eq. (4) can be written as follows.

$$E_{D,n} = m_e c^2 - \frac{1}{2} \left(\frac{1}{4\pi\epsilon_0} \right)^2 \frac{m_e e^4}{\hbar^2} \frac{1}{n^2} \quad (6a)$$

$$= m_e c^2 + E_n. \quad (6b)$$

Here, the D in E_D indicates the equation derived by Dirac. $E_{D,n}$ of Eq. (6) expresses the remaining amount of rest mass energy of the electron. Even if we place an electron at rest an infinite distance from its nucleus, the relativistic energy of the electron is not zero.

Taking these facts into account, the relativistic energy $E_{re,n}$ of the electron is defined as follows.⁸

$$E_{re,n} = m_e c^2 + E_n, \quad n=1,2,\dots \quad (7)$$

Also, the author has derived an energy-momentum relationship applicable to the electron in a hydrogen atom by referring to the reasoning Einstein used when deriving Eq. (2).⁸ That is,

$$E_{re,n}^2 + p_n^2 c^2 = (m_e c^2)^2. \quad (8)$$

In addition, Eq. (8) was verified in that paper.

III. DERIVATION OF ULTRA-LOW ENERGY LEVELS OF THE HYDROGEN ATOM

Dirac showed that the energy in Eq. (2) has the following positive and negative values.⁹

$$E = \pm c(m_e^2 c^2 + p^2)^{1/2}. \quad (9)$$

Here, the following inequality holds for the negative energy.

$$E < -m_e c^2. \quad (10)$$

If Eq. (8) is solved by following Dirac, the following solution is obtained.

$$E_{re,n} = \pm c(m_e^2 c^2 - p_n^2)^{1/2}. \quad (11)$$

The range of values that can be assumed by $E_{re,n}$, obtained from this equation, is as follows.

$$m_e c^2 > E_{re,n} > -m_e c^2. \quad (12)$$

For this reason, it is theoretically possible for an energy level satisfying the condition $0 \geq E_{re,n} > -m_e c^2$ to exist.

Incidentally, according to the virial theorem, the following relation holds between K and V :

$$\langle K \rangle = -\frac{1}{2} \langle V \rangle. \quad (13)$$

Here, K is the kinetic energy of the entire system, and V is the potential energy of the entire system.

The average time of K is equal to $-1/2$ the time average of V . Also, the sum of the time average K and the time average of the total mechanical energy E of the entire system becomes 0. That is,

$$\langle K \rangle + \langle E \rangle = 0. \quad (14)$$

Next, if Eqs. (13) and (14) are combined, the result is as follows:

$$\langle E \rangle = -\langle K \rangle = \frac{1}{2} \langle V \rangle. \quad (15)$$

Taking these facts into account, the author presented the following equation as an equation indicating the relationship between the rest mass energy and potential energy of the electron in the electrostatic field of the proton.¹⁰

$$V(r) = -\Delta m_e c^2. \quad (16)$$

According to this equation, the potential energy corresponds to the reduction in rest mass energy of the electron in the atom. Also, the potential energy of the electron can be expressed with the following equation:

$$V(r_n) = -\frac{1}{4\pi\epsilon_0} \frac{e^2}{r_n} \quad (17a)$$

$$= -\frac{\alpha\hbar c}{r_n}. \quad (17b)$$

Here, if Eq. (8) is rewritten taking into account Eq. (15), the following two equations are obtained.

$$\left(m_e c^2 - \frac{\alpha^2 m_e c^2}{2n^2}\right)^2 + p_n^2 c^2 = (m_e c^2)^2. \quad (18)$$

$$\left(m_e c^2 - \frac{\alpha\hbar c}{2r_n}\right)^2 + p_n^2 c^2 = (m_e c^2)^2. \quad (19)$$

The following p_n^2 can be found from Eq. (18).

$$p_n^2 = \frac{\alpha^2 m_e^2 c^2}{n^2} \left(1 - \frac{\alpha^2}{4n^2}\right). \quad (20)$$

Expanding and rearranging Eq. (19), it is possible to obtain the following quadratic equation for r_n :

$$p_n^2 r_n^2 - \alpha\hbar m_e c r_n + \frac{\alpha^2 \hbar^2}{4} = 0. \quad (21)$$

If the value of Eq. (20) is substituted here for p_n^2 , the following solution is obtained:

$$r_n = \frac{\alpha\hbar m_e c \pm (\alpha^2 \hbar^2 m_e^2 c^2 - p_n^2 \alpha^2 \hbar^2)^{1/2}}{2p_n^2} \quad (22a)$$

$$= \left[r_e \pm r_e \left(1 - \frac{\alpha^2}{2n^2}\right) \right] \frac{n^2}{2\alpha^2} \left(1 - \frac{\alpha^2}{4n^2}\right)^{-1}. \quad (22b)$$

Here, r_e is the classical electron radius as following:

$$r_e = \frac{1}{4\pi\epsilon_0} \frac{e^2}{m_e c^2}. \quad (23)$$

Here if r_n^+ is taken to be the larger of the two solutions obtained from Eq. (22), and r_n^- the smaller, then r_n^+ and r_n^- are as follows:

$$r_n^+ = \frac{n^2}{\alpha^2} r_e = n^2 a_B, \quad n=1,2,\dots \quad (24)$$

$$r_n^- = \frac{r_e}{4} \left(\frac{n^2}{n^2 - \alpha^2/4} \right), \quad n=1,2,\dots \quad (25)$$

Here, a_B is the Bohr radius. In Eq. (25) the radius approaches $r_e/4$ with $n \rightarrow \infty$. Therefore this paper predicts that $r_e/4$ will be the radius of the atomic nucleus (the proton).

Next, the energy is found when the radius is r_n^+ and r_n^- . Here, E_n in Eq. (3) can be written as follows if Eq. (15) and Eq. (17a) are taken into consideration.

$$E_n = -\frac{1}{2} \frac{1}{4\pi\epsilon_0} \frac{e^2}{r_n}. \quad (26)$$

Here, if E_n^+ is taken to be the energy obtained by substituting r_n^+ for r_n in Eq. (26), and similarly E_n^- is taken to be the energy obtained by substituting r_n^- for r_n , then E_n^+ and E_n^- are as follows:

$$E_n^+ = E_{B,n} = -\frac{\alpha^2 m_e c^2}{2n^2}, \quad n=1,2,\dots \quad (27)$$

$$E_n^- = -2m_e c^2 + \frac{\alpha^2 m_e c^2}{2n^2}, \quad n=1,2,\dots \quad (28)$$

If these energies are described on an absolute scale using $E_{re,n}$ defined in Eq. (7), the values are as follows:

$$E_{re,n}^+ = m_e c^2 - \frac{\alpha^2 m_e c^2}{2n^2}, \quad n=1,2,\dots \quad (29)$$

$$E_{re,n}^- = -m_e c^2 + \frac{\alpha^2 m_e c^2}{2n^2}, \quad n=1,2,\dots \quad (30)$$

If the above is indicated graphically, the result is as follows (see Fig. 1).

IV. PHENOMENA WHICH DEMONSTRATE THE EXISTENCE OF ULTRA-LOW ENERGY LEVELS

If it is assumed that the energy of an incident γ -ray is 1.022 MeV ($2m_e c^2$), then due to the effects of the Coulomb potential of atomic nucleus, the γ -ray may suddenly disappear and produce an electron-positron pair (electron-pair creation). The electron and positron pair, which absorbed all of the energy of the γ -ray, are produced, classically speaking, near $r_e/2$ from the center of the atomic nucleus.

If an electron exists at the E_n^- energy levels, then the energy E necessary for the electron to be excited and emitted outside of the shell is as follows due to Eq. (28):

$$E = 2m_e c^2 - \frac{\alpha^2 m_e c^2}{2n^2} \approx 2m_e c^2 = 1.022 \text{ MeV}. \quad (31)$$

Now, how can hydrogen atoms in this energy state be verified? Simply put, it can be predicted that if such an atom is irradiated with a 1.022 MeV photon, then a single electron will be emitted to the outside of the shell. However, this sort of phenomenon has not been

discovered. Actually, the $r=r_e/2$ point is a location where energetically $E_{re}=0$ and an electron-positron pair is created due to the vacuum absorbing the energy of this photon. Since the photon energy is consumed at that time, the photon cannot arrive at an electron in the E_n^- state. Therefore this paper looks at triplet production.

It is generally assumed that in triplet production, in which 2 electrons and 1 positron are created, electron-pair creation occurs not near the atomic nucleus, but near the electron in the outer shell orbital. A total of three particles are created in this case: one outer shell electron forming the atom, and a positron and electron created through pair production. However, in this model, $(1.022\text{MeV} - E_n^+)$ should be sufficient as the necessary photon energy. If an energy of 2.044MeV ($4m_e c^2$), is needed for triplet production, then the recoiled electron should be regarded as being at an ultra-low energy level. Thus this theory changes the existing interpretation of triplet production by taking into account the law of conservation of energy.

Now, consider the case where an incident γ -ray has the energy corresponding to the mass of 4 electrons (2.044 MeV). If this is discussed classically, the γ -ray can create an electron and positron near $r=r_e/2$ (see Fig.2).

Even if 1.022 MeV of energy is consumed in this pair creation, the γ -ray still has the energy of corresponding to the mass of 2 electrons (1.022MeV). If the γ -ray gives energy to an electron in the orbital near the proton, the electron will be excited and appear in free space. As a result, 2 electrons and 1 positron will appear in free space.

This paper points out that one of the two electrons which appears is an electron in the E_n^- state. A hydrogen atom in the E_n^- state will henceforth be called a "dark hydrogen atom" in this paper, and indicated as ${}_D\text{H}$ (where the D stands for "dark"). Also, the E_n^- in Eq. (28) will be indicated as ${}_D E_n^-$, and E_n^+ will be returned to the original symbol E_n . In addition, the $E_{re,n}^-$ in Eq. (30) will be expressed as ${}_D E_{re,n}$, and $E_{re,n}^+$ will be returned to the original symbol $E_{re,n}$.

Furthermore, the hydrogen molecule produced from ${}_D\text{H}$ will be called "dark hydrogen molecule", and indicated as ${}_D\text{H}_2$. In this paper, ${}_D\text{H}$ and ${}_D\text{H}_2$ will be grouped together and called either hydrogen at ultra-low energy levels, or "dark hydrogen". In the next section, this paper will examine whether dark hydrogen can be a DM candidate.

V. DISCUSSION

This section examines whether ${}_D\text{H}$ and ${}_D\text{H}_2$ have the six characteristics indicated in the introduction.

(1) Hydrogen atoms are the most common atoms in the universe. Therefore, if dark hydrogen exists, it is

natural to assume that it is present universally throughout space. Also, it is clear that it is electrically neutral. Therefore characteristics (1) and (2) are satisfied.

(2) The relativistic energy E_{re} of the electron forming H is about $\approx m_e c^2$ (however, $E_{re} < m_e c^2$). In contrast, the relativistic energy E_{re} of the electron forming ${}_D\text{H}$ is about $\approx -m_e c^2$ (however, $E_{re} > -m_e c^2$).

Also, as is clear from Eq. (30), $E_{re} < 0$ in the case of the ${}_D\text{H}$ electron, and therefore the mass of the electron forming ${}_D\text{H}$ has a negative value. ${}_D\text{H}$ is about $2m_e c^2$ lighter than H. However, ${}_D\text{H}$ has a far smaller radius than H. Now, if the classical radii of ${}_D\text{H}$ and H are compared, the results are as follows:

$$\frac{r_1^-}{r_1^+} = \frac{\alpha^2}{4 - \alpha^2} \quad (32a)$$

$$= 1.331 \times 10^{-5}. \quad (32b)$$

Therefore, dark hydrogen can achieve a state of far higher density than ordinary hydrogen. It is likely it will be observed as matter with high mass. Therefore, characteristic (3) can be regarded as satisfied.

(3) In atoms and molecules, electrons occupy orbitals starting from states with low energy. However, as shown by Eq. (14), there is a lower limit on the potential energy of an electron in a hydrogen atom. That is,

$$0 > V(r) > -m_e c^2. \quad (33)$$

When an electron in a hydrogen atom makes a transition from E_n^+ to E_n^- , the electron must have the energy of a photon emitted to the outside of the shell. However, the energy necessary for this transition does not remain at the electron if Eq. (33) is taken into consideration. Therefore, a transition from E_n^+ to E_n^- does not occur. In contrast, the transition from E_n^- to E_n^+ is regarded as possible. If a dark electron absorbs $2m_e c^2$ of energy from the outside, the electron in ${}_D\text{H}$ is excited, and it transitions to an orbital within the hydrogen atom, or is omitted outside of the shell [This satisfies characteristic (4)].

It is also likely that dark hydrogen satisfies characteristic (5) and (6). For the above reasons, this paper has determined that ${}_D\text{H}$ and ${}_D\text{H}_2$ can be DM candidates.

VI. CONCLUSION

This paper has shown it as possible to demonstrate the existence of ${}_D\text{H}$ by changing the interpretation of the phenomenon of triplet production. Also, the characteristics of ${}_D\text{H}$ and ${}_D\text{H}_2$ match with the six characteristics indicated in the introduction.

This paper predicts that, in addition to such dark hydrogen, diverse types of atoms and various types of molecules comprised of diverse types of atoms, can also be candidates for Dark Matter.

would like to express my thanks to the staff at Translation Services for their translation assistance. I wish to express my gratitude to Mr. H. Shimada for his figures.

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Fig.1 : The energy levels of the hydrogen atom predicted by classical quantum theory E_n^+ ($E_{re,n}^+$) and the new energy levels whose existence has been indicated by this paper E_n^- ($E_{re,n}^-$). Also, r is physical quantity with meaning within the scope of classical discussions. The region of physical vacuum where pair production is possible in the hydrogen atom is $r_e/2 \geq r > r_e/4$.

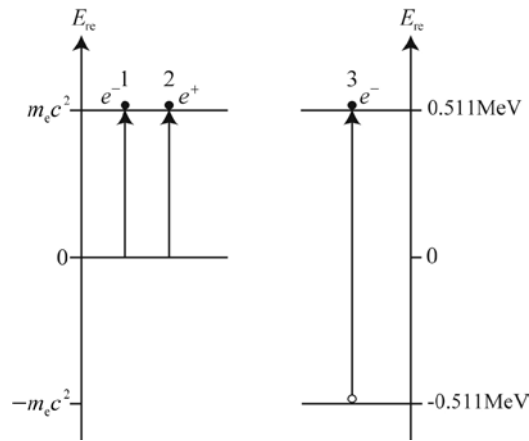


Fig. 2a

Fig. 2b

Fig. 2 : Interpretation of this paper regarding triplet production. From the perspective of this paper, this γ -ray will give 1.022 MeV of energy to the virtual particles at $r=r_e/2$, and an electron-positron pair will be created (Fig. 2a). When this γ -ray approaches closer to the atomic nuclear, and the electron in the orbital around the proton absorbs this energy, the electron will be excited and appear in free space (Fig. 2b). This paper points out that one of the two electrons which appears is an electron in the E_n^- state.



Optimization Control SK Box Manoeuvres for GEO Satellites using Electric Thrusters (OCSKBOX)

By Louardi Beroual & Djamel Benatia

University of Batna, Algeria

Abstract- The study presented in this paper deals with an optimization control station keeping box manoeuvres for geostationary satellites equipped with electric propulsion.

The station keeping box (SKBOX) represented the maximum permitted values of the excursion of the satellite in longitude and latitude. It can be represented as a pyramidal solid angle, whose vertex is at the centre of the earth, within which the satellite must remain at all times. In this work, the station keeping box is defined by the two half angles at the vertex, one within the plan of the equator (E-W width), and the other in the plan of the satellite meridian (N-S width).

A number of different techniques are available for the numerical solution of the station keeping box problem. In this work we will consider the direct method for solution of continuous optimal control problem. Simulation results have demonstrated that the spacecraft can be tightly controlled within station keeping box.

Keywords: geostationary satellites, SKBOX, box-limit, electric propulsion, specific impulse, quadratic programming.

GJSFR-A Classification : FOR Code: 290207



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I. INTRODUCTION

An astrodynamics orbital station-keeping is the orbital maneuvers made by thruster burns that are needed to keep a spacecraft in a particular assigned orbit.

For many Earth satellites the effects of the non-Keplerian forces, i.e. the deviations of the gravitational force of the Earth from that of a homogeneous sphere, gravitational forces from Sun/Moon, solar radiation pressure and air-drag must be counteracted.

The deviation of Earth's gravity field from that of a homogeneous sphere and gravitational forces from Sun/Moon will in general perturb the orbital plane [1]. For geostationary spacecraft the inclination change caused by the gravitational forces of Sun/Moon must be counteracted to a rather large expense of fuel, as the inclination should be kept sufficiently small for the spacecraft to be tracked by a non-steerable antenna [1,2].

Solar radiation pressure will in general perturb the eccentricity (i.e. the eccentricity vector) [3]. For some missions this must be actively counter-acted with manoeuvres, for geostationary spacecraft the

eccentricity must be kept sufficiently small for a spacecraft to be tracked with a non-steerable antenna. Also for Earth observation spacecraft for which a very repetitive orbit with a fixed ground track is desirable, the eccentricity vector should be kept as fixed as possible. A large part of this compensation can be done by using a frozen orbit design, but for the fine control manoeuvres with thrusters are needed [4].

Electric propulsion engines are more efficient than chemical ones: they require significantly less propellant to produce the same overall effect, for example a specific increase in spacecraft velocity. The propellant is ejected up to 20 times faster than from chemically-based thrusters, and, thus the same propelling force is obtained with a log less propellant, the forces EP produces can be applied continuously for very long periods-months or even years.

To maintain the satellite within box, orbit corrections are achieved by applying velocity impulses to the satellite at a point in the orbit. These impulses are generated by activating the thrusters that are mounted on the satellite as part of the propulsion subsystem.

The tool developed in the frame of this paper is based on numerical optimization techniques and uses a thrusters-based model of the satellite to take directly into account the activity of each thruster used for the control of the satellite on the optimization process. This paper presents the tool design and the main principle of the optimization algorithm. Usually, control strategies consider satellites as a point. The present work includes the mathematical definition and the satellite model that allow considering it as a system. The results of some simulations and their practical applications are presented.

II. MATHEMATICAL MODELING

a) Coordinate Frames

The coordinate system used in this work for describing the perturbing forces is the satellite based Radial Tangent Normal (RTN) coordinate system with orthonormal basis $\vec{R}\vec{T}\vec{N}$. The $\vec{R}$ axis is defined as always pointing from the Earth's center along the radius vector toward the satellite, The $\vec{N}$ axis is normal to the orbit plane with direction of the satellite angular momentum vector, The $\vec{T}$ axis is perpendicular to $\vec{R}$ in the orbit plane

Author ^α : Department of Electrical Engineering, University of Batna, Rue Chahid Boukhrouf, CUBI, Batna, 05000, Algeria.
e-mail: louardi_b@yahoo.fr

and with the direction toward the satellite movement. It completes, with the unit vectors $\vec{R}$ and $\vec{N}$, a right-handed orthogonal basis (see Figure 1).

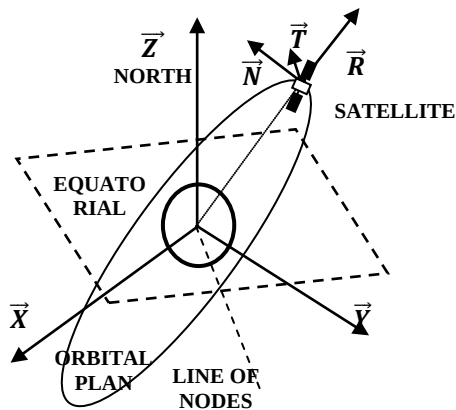


Figure 1 : Coordinate frames

In the following, a generic acceleration vector $\vec{u}$ induced from propulsive force acting on the satellite will be expressed as

$$\vec{u} = u_R \vec{R} + u_T \vec{T} + u_N \vec{N} \quad (1)$$

Where u_R, u_T, u_N are the acceleration components along the radial, tangential and normal directions.

b) Orbit elements

A total of six independent parameters are required to describe the motion of a satellite around the earth [3,4]. Two of these elements, semi-major axis a

and eccentricity e describe the form of the orbit, one element the mean anomaly M defines the position of satellite along the orbit, the three others, the right ascension Ω , inclination i and argument of perigee ω define the orientation of the orbit in space. Given these six elements, it is always possible to uniquely calculate the position and velocity vector (see figure. 2).

In many application, satellite orbits are chosen to be near-circular, to provide a constant distance from the surface of the Earth or a constant relative velocity. Typical examples are low-altitude remote sensing satellite or geostationary satellite.

While there is no inherent difficulty in calculating position and velocity from known orbital elements with e and i close to zero, the reverse task may cause practical and numerical problems. These problems are due to singularities arising from the definition of some of the classical orbital elements. The argument of perigee ω , for example, is not a meaningful orbital element for small eccentricities, since the perigee itself is not well defined for an almost circular orbit. Similar consideration apply to small inclination i where the line of node is no longer well defined and where the equations for Ω become singular. Several attempts have therefore been made to substitute other parameter for the classical keplerian elements. These elements are usually referred to as non-singular, regular or equinoctial elements [4].

The satellite orbit plane is defined thanks to the component of the inclination vectors (with modulus $\tan(i/2)$) directed along the line of nodes and pointing towards the ascending node.

$$\vec{i} = [p \ q]^T = 2 \tan\left(\frac{i}{2}\right) [\cos\Omega \ \sin\Omega]^T \cong [\cos\Omega \ \sin\Omega]^T, i \rightarrow 0 \quad (2)$$

The satellite trajectory on its orbit is defined to the semi major axis a , and supposing the parameters Ω and ω in the same plane, to the component of the

eccentricity vector directed along the line of apsis and pointing towards the perigee.

$$\vec{e} = [h \ k]^T = e [\cos(\Omega + \omega) \ \sin(\Omega + \omega)]^T = e [\cos\tilde{\omega} \ \sin\tilde{\omega}]^T \quad (3)$$

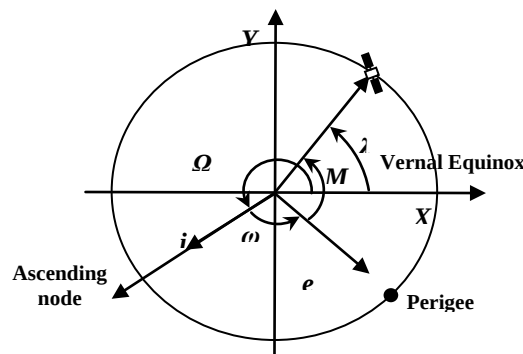


Figure 2 : Orbit elements

Finally, the position of the satellite along its orbit is represented by the mean longitude

$$l = \tilde{\omega} + M - \theta(t) \quad (4)$$

Where Θ is the Greenwich sidereal angle.

c) Dynamics for a GEO satellite

The motion of GEO satellite can be described by the rate of change of the equinoctial orbital parameters under the influence of the forces acting on the satellite. The geostationary dynamics results in the following nonlinear time varying system

$$\dot{x}(t) = \mathcal{K}(x(t)) + \mathcal{L}(t, x(t)) + \mathcal{G}(t, x(t))u(t) \quad (5)$$

Where

$$x = [a \ p \ q \ h \ k \ l]^T \quad (6)$$

$$u = [u_R \ u_T \ u_N]^T \quad (7)$$

And the functions, $\mathcal{K}$, $\mathcal{L}$ and $\mathcal{G}$ are the variation contribution to the equinoctial elements coming respectively from the Kipler's, Lagrange's and Gausse planetary equations [6] and [7].

$\mathcal{K}$: Describes the satellite motion under the effect of the gravitational attraction of the earth considered with takes into account the effect of the natural perturbing forces.

$\mathcal{L}$: Take into account the effect of the natural perturbing forces.

$\mathcal{G}$: Given by the acceleration by thrusts.

The translation of nonlinear model (equation 5) into the linear model we use the Taylor series up to first order around the nominal operating points

$$x_0 = [a_0 \ 0 \ 0 \ 0 \ 0 \ l_0]^T \quad (8)$$

$$u_0 = [0 \ 0 \ 0]^T \quad (9)$$

We obtain

$$\dot{x}(t) = \mathcal{A}(t)x(t) + \mathcal{B}(t)u(t) + \mathcal{D}(t) \quad (10)$$

Where

$$\bar{x} = x - x_0 \text{ and } \bar{u} = u - u_0 \quad (11)$$

The $\mathcal{A}(t)$ and $\mathcal{D}(t)$ matrices turn out to be time varying because of the presence of periodic terms with periods equal to multiples of the periodicities of the earth, sun and Moon motion relative to the satellite.

The matrix $\mathcal{B}(t)$ is a periodic function with period equal to 24 hours.

$$\mathcal{B}(t) = \frac{1}{v_0} \begin{bmatrix} 0 & 2a_0 & 0 \\ 0 & 0 & \frac{1}{2}\sin\psi_0 \\ 0 & 0 & \frac{1}{2}\cos\psi_0 \\ -\cos\psi_0 & 2\sin\psi_0 & 0 \\ \sin\psi_0 & 2\cos\psi_0 & 0 \\ -2 & 0 & 0 \end{bmatrix} \quad (12)$$

Where

v_0 station keeping velocity equal to $\sqrt{\frac{\mu}{a_0}}$, μ is the earth gravitational coefficient.

And

$$\psi_0 = l_0 + \theta(t) \quad (13)$$

The GEO orbits are characterized by very small values of eccentricity e and inclination i . the longitude and latitude can be defined

$$\lambda = l + 2h \sin(l + \theta) - 2k \cos(l + \theta) \quad (14)$$

$$\varphi = 2p \sin(\lambda + \theta) - 2q \cos(\lambda + \theta) \quad (15)$$

We denote y the spacecraft position vector, which can be considered as the output variable of the nonlinear model

$$y = f(x, t) \quad (16)$$

The output equation into its Taylor series up to the first order around x_0 [8], we get the output equation of the linear time varying system (eq.10)

$$y = C(t)\bar{x} \quad (17)$$

Where

$$y = [\lambda - l_0 \ \varphi]^T \quad (18)$$

And

$$C(t) = \begin{bmatrix} 0 & -a_0 \cos\psi_0 & -a_0 \sin\psi_0 & 0 & 0 & 0 \\ 1 & -2\cos\psi_0 & 2\sin\psi_0 & 0 & 0 & 1 \\ 0 & 0 & 0 & 0 & -2\cos\psi_0 & 2\sin\psi_0 \end{bmatrix} \quad (19)$$

III. STATION KEEPING BOX PROBLEM FORMULATION

The station keeping box represents the maximum permitted values of the excursion of satellite in longitude λ and latitude φ . The SK box can be considered as pyramidal solid angle, whose vertex is at centre of the earth.

Is defined by the two half angles of vertex, one within plan of equator $E-W$ width $2\lambda_{max}$ and other in the plan satellite meridian $N-S$ width $2\varphi_{max}$ (see Figure. 3).

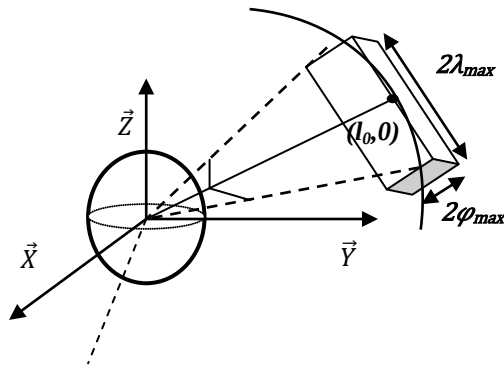


Figure 3 : Station keeping box

The station keeping problem can be formulated as constrained linear quadratic continuous time optimal control problem [9]. Given the linear model equation (10) and equation (17) with initial condition $\bar{\mathbf{X}}(t_i) = \bar{\mathbf{X}}_i$, the problem is to find the control optimal $\bar{\mathbf{u}}_{opt}(t)$ over a finite time horizon $t_f - t_i$ the minimize the criterion

$$\Gamma = \begin{bmatrix} -\sin\gamma \cos\sigma & -\sin\gamma \cos\sigma & -\sin\gamma \cos\sigma & -\sin\gamma \cos\sigma \\ \sin\gamma \sin\sigma & \sin\gamma \sin\sigma & \sin\gamma \sin\sigma & \sin\gamma \sin\sigma \\ -\cos\gamma & -\cos\gamma & \cos\gamma & \cos\gamma \end{bmatrix} \quad (24)$$

And $\mathbf{F}$ is the thrust vector of the thruster system

$$0 \leq F \leq F_{max} \quad (25)$$

We defined the constraints on the control variable by

$$\frac{-F_{max}}{m} \leq \Gamma(\Gamma\Gamma^T)^{-1}\mathbf{u} \leq \frac{F_{max}}{m} \quad (26)$$

IV. NUMERICAL SOLUTION OF THE PROBLEM

A number of different techniques are available for the numerical solution of the station keeping box problem. In this work we will consider the direct method for solution of continuous optimal control problem, the idea behind direct method is to discrete the control time history and/or state variable history [10,11].

$$\mathcal{L}(t) = \begin{bmatrix} 0 & 0 & 0 & 2\cos\psi_0 & -2\sin\psi_0 & -1 \\ 0 & 0 & 0 & 2\cos\psi_0 & -2\sin\psi_0 & -\frac{3}{2} \\ -\frac{3}{2a_0} & 0 & 0 & 2\sin\psi_0 & 2\cos\psi_0 & 0 \\ 0 & 2\cos\psi_0 & -2\sin\psi_0 & 0 & 0 & 0 \\ 0 & 2\sin\psi_0 & 2\cos\psi_0 & 0 & 0 & -2\sin\psi_0 \\ -\frac{2}{a_0} & 0 & 0 & 2\sin\psi_0 & 2\cos\psi_0 & 0 \end{bmatrix} \quad (28)$$

The linear system (equation 10) can be written in the new state variables $\tilde{\mathbf{x}}$ as

$$\dot{\tilde{\mathbf{x}}}(t) = \tilde{\mathbf{A}}(t)\tilde{\mathbf{x}}(t) + \tilde{\mathbf{B}}(t)\mathbf{u}(t) + \tilde{\mathbf{D}}(t) \quad (29)$$

And we can write the control variable of the linear dynamics as a function of the state variable, so

$$\mathbf{u}(t) = \tilde{\mathbf{B}}^{-1}\tilde{\mathbf{x}}(t) - \tilde{\mathbf{B}}^{-1}\tilde{\mathbf{A}}(t)\tilde{\mathbf{x}}(t) - \tilde{\mathbf{B}}^{-1}\tilde{\mathbf{D}}(t) \quad (30)$$

Where

$$J = \frac{1}{2} \int_{t_i}^{t_f} (\mathbf{y}^T(t) \mathbf{Q}(t) \mathbf{y}(t) + \mathbf{u}^T(t) \mathbf{R}(t) \mathbf{u}(t)) dt \quad (20)$$

Subject to the conditions

$$-y_{max} \leq y \leq y_{max} \quad (21)$$

Where

$$\mathbf{y}_{max} = [\lambda_{max} \quad \phi_{max}] \quad (22)$$

The thruster accelerations are defined as control laws in the optimization problems. These control variables can thus take at any time any value comprised between zero and the maximum thruster acceleration, in general with j thruster, we can write the control vectors in RTN fram as

$$\mathbf{u}_j = \frac{1}{m} \Gamma \mathbf{F}_j \quad (23)$$

Where m is the spacecraft and Γ is the thruster system configuration matrix can be defined for a satellite equipped with four electric thrusters as

In this technique, the control inputs have to be written explicitly as function of the state and its rate of change so that bounds on the control variables have translated in bounds on the attainable rates of change of the state variable [12].

The linear model (equation 10) can be written in different form, characterized by matrix $\mathbf{B}$. to this purpose, we can use the Lyapunov transformation [9,13], in the state space defined as

$$\tilde{\mathbf{x}} = \mathcal{L}(t)\mathbf{x} \quad (27)$$

Where

$$\tilde{\mathbf{B}}^{-1} = [\mathcal{L}(t)\mathbf{B}(t)]^{-1} \quad (31)$$

We obtained

$$\mathbf{u}(t) = \mathbf{M}\tilde{\mathbf{x}}(t) - \mathbf{M}\tilde{\mathbf{A}}(t)\tilde{\mathbf{x}}(t) - \mathbf{M}\tilde{\mathbf{D}}(t) \quad (32)$$

Where

$$\mathbf{M} = \tilde{\mathbf{B}}^{-1} = \begin{bmatrix} 0 & 0 & 1 & 0 & 0 & 0 \\ 0 & 0 & 0 & 1 & 0 & 0 \\ 0 & 0 & 0 & 0 & 1 & 0 \end{bmatrix} \quad (33)$$

And can be write the state variable as

$$\dot{\tilde{x}}(t) = \tilde{A}(t)\tilde{x}(t) + \tilde{D}(t) \quad (34)$$

And the output equation given by

$$y(t) = \tilde{C}(t)\tilde{x}(t) \quad (35)$$

Where

$$\tilde{C}(t) = C(t)\mathcal{L}^{-1}(t) \quad (36)$$

The station keeping problem formulated in the previous section as a constrained continuous time optimal control problem can be translated in a quadratic programming problem with constraint only on the state variables.

Above a finite time horizon $t_f - t_i$ discretized in N intervals of length equal h each. The control optimal $u^{opt}(t)$ is taken constant equal to

$$u_k^{opt} \quad \text{With } k=0, 1, \dots, N-1 \quad (37)$$

The problem is consist in finding the optimal sequence $\tilde{x}_k^{opt}$ that minimized the criterion

$$J = \frac{1}{2} \sum_{k=1}^N y_k^T Q_k y_k + \frac{1}{2} \sum_{k=0}^{N-1} u_k^T R_k u_k \quad (38)$$

With

$$y_k = \tilde{C}_k \tilde{x}_k \quad (39)$$

$$u_k = M \frac{\tilde{x}_{k+1} - \tilde{x}_k}{h} - M \tilde{A}_k \frac{\tilde{x}_{k+1} - \tilde{x}_k}{2} - M \tilde{D}_k \quad (40)$$

Subject

- The output variable $y(t)$

$$-y_{max} \leq y_k \leq y_{max} \quad (41)$$

- The control variable $u(t)$

$$\frac{-F_{max}}{m} \leq \Gamma(\Gamma\Gamma^T)^{-1} u_k \leq \frac{F_{max}}{m} \quad (42)$$

- The auxiliary state variable $\tilde{x}(t)$

$$\frac{\tilde{x}_{k+1} - \tilde{x}_k}{h} = \tilde{A}_k \frac{\tilde{x}_{k+1} + \tilde{x}_k}{2} + \tilde{D}_k \quad (43)$$

A step-by-step walkthrough of the algorithm is as follows:

Step1: Formulation of SKBOX problem

- Fixed the finite time horizon $t_f - t_i = 1$ day.
- The weighting matrices are equal to: $R = I_{2 \times 2}$ and $Q = I_{2 \times 2}$.
- Fixed the initial orbital elements vector $x(t_i)$.

Step2: Finding the optimal solution with minimized the criterion J with a discretization step of length $h=0.01$ day.

Step3: Obtained the optimal control with equation (40).

Step4: Finding the output variable with equation (39).

Step5: Repeat the previous steps for 1yaer.

V. NUMERICAL SIMULATION

The initial orbital elements for this simulation can be found in Table 1.

Table I : Initial Orbital Elements

Orbital parameter	Value
Semi major axis(km)	42166.279
Right ascension of A.N (°)	82
Eccentricity	0.0000778
Inclination (°)	0.002044
Argument of perigee (°)	315.67725
Mean anomaly (°)	324.34109

The satellite is considered in this simulation is equipped with four thrusters mounted on its anti nadir face (see Figure 4).the configuration of the force vector is

$$F = [F_1 \ F_2 \ F_3 \ F_4] \quad (42)$$

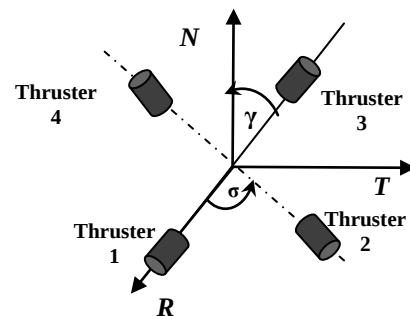


Figure 4 : Configuration force vector

The characteristics for this satellite can be found in Table 2.

Table II : Characteristics of Satellite

Characteristics of satellite	Value
Spacecraft masse (kg)	4500
Can angles of thruster (°)	50
Slaw angles of thruster (°)	15
Maximum force modulus (N)	0.17
Specific impulse (s).	3800s

The objective is to determine the set of manoeuvres to be executed in order to keep the satellite in a latitude and longitude box centered at the station longitude $l_0=10\text{deg}$ with $\lambda_{\max}=0.01\text{deg}$ and $\varphi_{\max}=0.01\text{deg}$.

Figure 5, Figure 6 and Figure 7 illustrates the historical time of the optimal acceleration control components in RTN frame for one year.

Negative and positive values of optimal acceleration radial allow maintain the satellite in a latitude window equal to $]-0.01^\circ, 0.01^\circ[$, positive values of optimal tangential acceleration fix the satellite into a longitude interval $]-0.01^\circ, 0.01^\circ[$, and the variation of

optimal acceleration normal centered the satellite in the box.

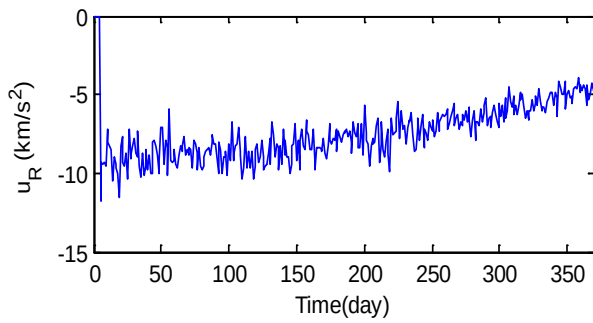


Figure 5 : Optimal acceleration radial control for one year

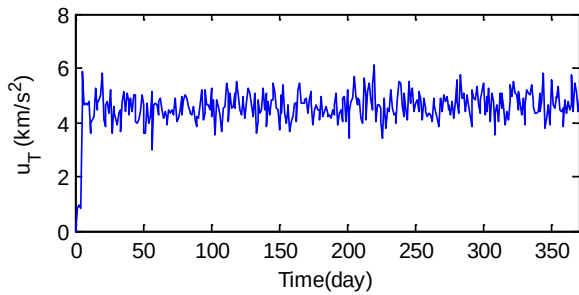


Figure 6 : Optimal acceleration tangential control for one year

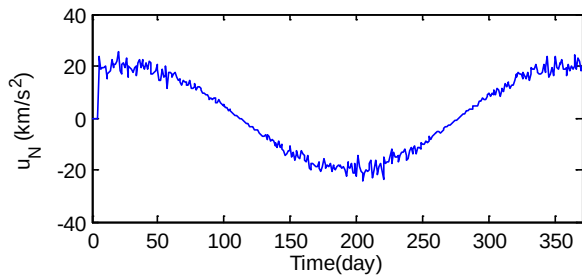


Figure 7 : Optimal acceleration normal control for one year

Figure 8 illustrates the evolution of the three components of propulsive force in *RTN* frame for one year with the maximum force modulus $F_{max}=0.17N$.

The variation in propulsive Force allows producing the optimal acceleration to control the satellite in latitude and longitude box. The thruster forces values are lower than maximum propulsive forces.

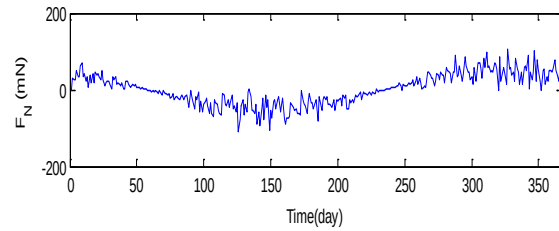
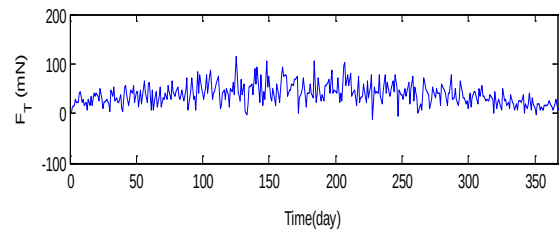
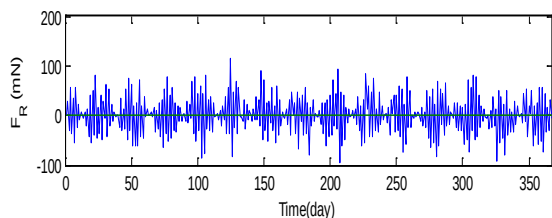


Figure 8 : Propulsive force for one year in RTN frame

Figure 9 Shows the controlled and uncontrolled manoeuvres time histories of the latitude for the station keeping box in condition $\pm 0.01^\circ$. In *SKBox* no-controlled, the variation in latitude value is not fixing in the box. For this problem the *SKBox* controlled is used for fixing the variation in latitude into the box.

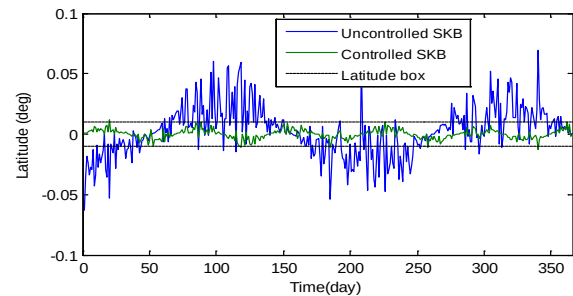


Figure 9 : Time histories of the latitude for one year

Figure 10 illustrates the controlled and uncontrolled manoeuvres time histories of the true longitude for the station keeping box in condition $\pm 0.01^\circ$. In *SKBox* no-controlled, the variation in longitude value is not fixing in the box. For this problem the *SKBox* controlled is used for fixing the variation in longitude into the box.

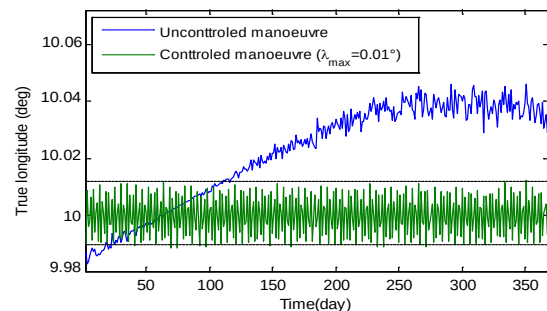


Figure 10 : Time histories of true longitude for one year

VI. CONCLUSION

In this paper a new method for station keeping box of geostationary satellite equipped with electric propulsion has been developed, we considered a novel approach based on direct method for solution of continues optimal control. Using this method, satellite position can be directly controlled based on the optimal acceleration for thruster, simulation results have demonstrated that the satellite can be tightly controlled in the station keeping box.

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Oscillating Theory of the Universe

By Koijam Manihar Singh, Kangujam Priyokumar Singh & Mukunda Dewri

Bodoland University, India

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Keywords: *dark energy, oscillating universe, scalar field, radiation/matter-dominated.*

GJSFR-A Classification : *MSC 2010: 83F05*



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Oscillating Theory of the Universe

Koijam Manihar Singh ^α, Kangujam Priyokumar Singh ^ο & Mukunda Dewri ^ρ

Abstract- We propose here a new theory of the cosmic evolution of the universe, giving its name as the "oscillating theory of the universe". In this theory we have an endless universe having different epochs in its continuing endless process, an epoch being equal to one complete oscillation starting from a big bang like event and undergoing through different phases until it arrives at a pseudo crunch to be followed by another big bang like event, of course, after a bounce. This moment of time is the end of an epoch and the beginning of another new epoch, which happens after a complete oscillation. In this oscillating theory the length of an epoch need not be equal to the length of another epoch. And interestingly, it is found that there exists the significance of negative time at some point during the cosmic evolution of the universe.

Keywords: dark energy, oscillating universe, scalar field, radiation/matter-dominated.

I. INTRODUCTION

There have been some theories and models of the universe, namely, standard big bang theory and inflationary models [Guth(1981), Albrecht and Steinhardt(1982), Linde (1982)], steady state theory [Hoyle and Narlikar (1964)], Brans-Dicke theory [Brans and Dicke (1961)], Dirac cosmology [Dirac (1973)], pre-big bang model [Gasperini and Veneziano (1993), Gasperini et al.(1997), Gasperini and Veneziano (2003)], ekpyrotic model [Khoury et al. (2001), Khoury et al. (2002a), Khoury et al. (2002b)], Horava-Witten heterotic M-theory [Horava and Witten (1996), Lukas et al. (1999a), Lukas et al. (1999b)], cyclic theory of the universe [Steinhardt and Turok (2002a), Steinhardt and Turok (2002b)], and the new ekpyrotic model [Buchbinder et al.(2007a), Buchbinder et al. (2007b), Buchbinder et al.(2008), Creminelli and Senatore (2007)]. In this paper, we attempt to introduce another new theory of the universe in which the universe undergoes an endless series of oscillations forming different epochs, and we name it as the "oscillating theory of the universe". Our model differs from oscillatory model introduced earlier based on a closed universe; rather our universe is flat and infinite. Our universe no longer passes through a

singularity in which the energy and the temperature diverge, rather the density and the temperature remain finite at the transition. Tolman (1934) pointed out that entropy produced during one cycle would be added to the entropy produced in the next, causing each cycle to be longer than the one before it. But his assumption is not necessarily true in our case. One oscillation may be longer or shorter than the next one as in our case the accelerated expansion, caused by the dark energy, is already diluting almost all the entropy, black holes and other debris accompanied with our universe which was produced in the previous epoch.

II. A BRIEF SKETCH OF THE OSCILLATING THEORY

In the proposed oscillating theory of the universe we consider the evolution of a scalar field ϕ coupled to our universe with gravity. In this theory one epoch begins with a big bang, after which the inflationary phase comes with a high rate of expansion. After a short period a stage is reached when the scalar field ϕ becomes nearly fixed during which the universe undergoes radiation-dominated and matter-dominated phases. Next during this process the universe reaches a stage when the potential energy of the scalar field begins to dominate giving a period of accelerated expansion which is the present age of the universe. During this period of accelerated expansion, the black holes, matter, radiation, all debris, neutron stars, neutrinos will be diluted away making the universe empty, smooth and flat. During this period the slope of the potential $V(\phi)$ causes the scalar field ϕ to roll down in the negative direction and the accelerated expansion will continue during which the potential energy almost drops to zero. At this point the universe will be dominated by the kinetic energy of the scalar field ϕ . But again, this kinetic energy will be damped away by the expansion of the universe. Then the moment comes when the total energy consisting of the kinetic energy and negative potential energy almost becomes zero at which moment the universe seems to be static momentarily. Then the scalar field starts to roll back towards $-\infty$ and the universe begins to contract. Now the kinetic energy of ϕ grows which means that the gravitational energy is being converted to the kinetic energy of ϕ . Thus the kinetic energy becomes increasingly dominant and the scalar field diverges as the radius of the universe tends to zero. Then a bounce follows and radiation is produced and the universe is beginning to expand. Though, for some time, the kinetic

Author α : Department of Mathematics, NIT Manipur, Imphal, PIN-795001, Manipur, India. e-mail: drmanihar@rediffmail.com

Author σ : Department of Mathematical Sciences, Bodoland University, Rangalikhata, Kokrajhar, B.T.C, Assam, PIN-783370, India. e-mail: pk_mathematics@yahoo.co.in

Author ρ : Department of Mathematical Sciences, Bodoland University, Rangalikhata, Kokrajhar, B.T.C, Assam, PIN-783370, India. e-mail: dewri11@gmail.com

energy density of the scalar field continues to dominate over the radiation, soon after the universe becomes radiation-dominated, and it undergoes a radiation-dominated phase, and then followed by a matter-dominated phase. During this phase the motion of the scalar field is rapidly damping, and it remains with almost reaching its maximum value. Then after a period of some billion years the potential energy of the scalar field begins to dominate, still ϕ rolling towards $-\infty$. As the scalar field rolls towards $-\infty$ the expansion of the universe begins diluting all the debris, the black holes, neutron stars etc. and flattening and making smooth the universe again. Then when all the energy of the scalar field is used up the universe seems to have a momentary pause, after which it (the universe) starts contracting leading to a pseudo crunch. Then after a bounce it goes for a big bang starting anew the process of evolution in the positive direction again. Thus the

universe goes on in an endless continuing process, where there is the significance of negative time in the evolution process of the universe, at least during the period when $\phi \rightarrow -\infty$. Here the periods of oscillation may not be necessarily equal.

III. AN OSCILLATING MODEL OF THE UNIVERSE

Here, as an illustration, we take up an oscillating universe. The line element considered is that of the flat and homogeneous Robertson-Walker metric

$$ds^2 = -dt^2 + a^2(t)[dr^2 + r^2(d\theta^2 + \sin^2\theta d\phi^2)] \quad (1)$$

where a is the scale factor. And the study of our model is taken up using the action S which describes the gravity, the scalar field ϕ , and the matter and radiation fluids in the form

$$S = \int d^4x \sqrt{-g} \left[\frac{1}{16\pi G} R - \frac{1}{2}(\partial\phi)^2 - V(\phi) + \psi^4(\phi)(\rho_r + \rho_m) \right], \quad (2)$$

where R is the Ricci scalar and $V(\phi)$ is the scalar potential. And it is such that the coupling $\psi^4(\phi)$ between the scalar field ϕ and the radiation (ρ_r) and the matter (ρ_m) densities causes the densities to remain finite at the moment of transition from the big crunch to the big bang event.

From (1) and (2) we get the field equations

$$\left(\frac{\dot{a}}{a}\right)^2 = \frac{8\pi G}{3} \left(\frac{1}{2}\dot{\phi}^2 + V + \psi^4\rho_r + \psi^4\rho_m \right) \quad (3)$$

$$\frac{\ddot{a}}{a} = -\frac{8\pi G}{3} \left(\dot{\phi}^2 - V + \psi^4\rho_r + \frac{1}{2}\psi^4\rho_m \right) \quad (4)$$

Here the scalar field ϕ satisfies the equation

$$\ddot{\phi} + 3H\dot{\phi} = -\frac{dV}{d\phi} - \frac{d\psi}{d\phi}\psi^3\rho_m \quad (5)$$

where

$$H = \frac{\dot{a}}{a} \quad (6)$$

is the Hubble constant.

For radiation or matter the equation of fluid motion is given by

$$a_\psi \frac{d\rho_\alpha}{da} = a \frac{d\rho_\alpha}{da} + \frac{\psi}{a} \frac{\partial \rho_\alpha}{\partial \phi} = -3(\rho_\alpha + p_\alpha) \quad (7)$$

where $\alpha = r, m$.

Also, $a_\psi = a\psi(\phi)$, and p_α is the pressure for the portion of the fluid whose energy density is ρ_α . Here assumption is implicitly made that radiation and matter couple to $\psi^2(\phi)g_{ij}$ (having scale factor a_ψ) rather than the Einstein metric g_{ij} alone (with scale factor a).

In our case we take the potential of the scalar field in the form

$$V(\phi) = V_0(1 - e^{-b\phi})f(\phi), \quad (8)$$

where $f(\phi)$ is a function such that $V(\phi) \rightarrow 0$ as $\phi \rightarrow -\infty$. With such a choice we can attain a realistic astrophysical situation by taking $b \geq 10$ and making today's dark energy density which is roughly $6 \times 10^{-30} \text{ gm/cm}^3$ equal to V_0 .

Regarding the coupling $\psi(\phi)$, we take

$$\psi(\phi) \sim e^{-\phi/\sqrt{6}} \text{ as } \phi \rightarrow -\infty \quad (9)$$

Here we choose $\psi(\phi)$ such that a_ψ and the matter and radiation densities are finite at $a = 0$ during the process of evolution. For large b , the potential is such that $\frac{V''}{V} \gg 1$ for $\phi_{\min} < \phi < 0$. And in this region the constant term is seen to be irrelevant and we may take $-V_0 e^{-b\phi}$ for V . Considering the motion of ϕ back and forth across the potential regime we find a solution to study the moments (periods) before and after the bounce. Over this region $-V_0 e^{-b\phi}$ can represent potential V , and a corresponding simple scaling solution is

$$a(t) = t^p, \quad (10)$$

$$V = -V_0 e^{-b\phi} = \frac{-p(1-3p)}{t^2}, \quad (11)$$

$$p = \frac{2}{c^2}, \quad (12)$$

which corresponds to an expanding or contracting universe according to t is positive or negative. And here $t = 0$ is chosen to correspond to a bounce. Thus at this particular juncture first t is negative which corresponds to a contracting universe leading to a pseudo big

crunch; then at $t = 0$ there is a bounce and just after that t is positive which corresponds to an expanding universe with a big bang. Thus at $t = 0$ one epoch of oscillation is completed, and from next moment onwards another epoch begins with a big bang.

IV. CONCLUSION

With the different arguments and reasons given above and observance of the continuity of the universe with such a theory and the results obtained here we have good reason to think of the oscillating theory as a very much viable new theory of the universe. Considering a model universe with a particular scalar field we have successfully shown that our universe can have different epochs of oscillation, one epochs consisting as usual of the radiation-dominated, matter-dominated and dark energy dominated phases. The different phases have been shown explicitly in our model. Finally we arrive at a juncture where an epoch of oscillation ends and after a bounce another epoch begins with a big bang. During each epoch there is a period of time when $\phi \rightarrow -\infty$ during which there is an implication for the existence of negative time; and such a similar situation arises also at the transit period from one epoch to another epoch. In this way the evolution of our universe is an endless process containing different epochs, each epoch comprising of a complete oscillation as we proposed.

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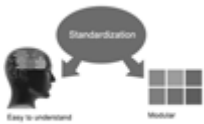
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Abstract:

The summary should be two hundred words or less. It should briefly and clearly explain the key findings reported in the manuscript-- must have precise statistics. It should not have abnormal acronyms or abbreviations. It should be logical in itself. Shun citing references at this point.

An abstract is a brief distinct paragraph summary of finished work or work in development. In a minute or less a reviewer can be taught the foundation behind the study, common approach to the problem, relevant results, and significant conclusions or new questions.

Write your summary when your paper is completed because how can you write the summary of anything which is not yet written? Wealth of terminology is very essential in abstract. Yet, use comprehensive sentences and do not let go readability for briefness. You can maintain it succinct by phrasing sentences so that they provide more than lone rationale. The author can at this moment go straight to shortening the outcome. Sum up the study, with the subsequent elements in any summary. Try to maintain the initial two items to no more than one ruling each.

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- Fundamental goal
- To the point depiction of the research
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Approach:

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Approach:

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Approach:

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What to keep away from

- Resources and methods are not a set of information.
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- Sum up your conclusion in text and demonstrate them, if suitable, with figures and tables.
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- Present a background, such as by describing the question that was addressed by creation an exacting study.
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- Recommendations for detailed papers will offer supplementary suggestions.

Approach:

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References	Complete and correct format, well organized	Beside the point, Incomplete	Wrong format and structuring



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